

principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office. The AMOC approval letter must specifically reference this AD.

(2) Contacting the Manufacturer: For any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; or the European Aviation Safety Agency (EASA); or Airbus's EASA Design Organization Approval (DOA). If approved by the DOA, the approval must include the DOA-authorized signature.

(i) Related Information

Refer to Mandatory Continuing Airworthiness Information (MCAI) EASA Airworthiness Directive 2014-0191, dated August 29, 2014, for related information. This MCAI may be found in the AD docket on the Internet at <http://www.regulations.gov/>#!documentDetail;D=FAA-2015-0086-0003.

(j) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless this AD specifies otherwise.

(i) Airbus Service Bulletin A310-71-2038, including Appendices 01 and 02, dated April 8, 2014.

(ii) Reserved.

(3) For service information identified in this AD, contact Airbus SAS, Airworthiness Office—EAW, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; email account.airworth-eas@airbus.com; Internet <http://www.airbus.com>.

(4) You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on July 2, 2015.

Jeffrey E. Duven,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2015-17202 Filed 7-15-15; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2006-23706; Directorate Identifier 2006-NE-03-AD; Amendment 39-18177; AD 2015-12-04]

RIN 2120-AA64

Airworthiness Directives; Honeywell International Inc. Turboprop Engines

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule; correction.

SUMMARY: The FAA is correcting an airworthiness directive (AD) that published in the **Federal Register**. That AD applies to all Honeywell International Inc. TPE331-1, -2, -2UA, -3U, -3UW, -5, -5A, -5AB, -5B, -6, -6A, -10, -10AV, -10GP, -10GT, -10P, -10R, -10T, -10U, -10UA, -10UF, -10UG, -10UGR, -10UR, -11U, -12JR, -12UA, -12UAR, and -12UHR turboprop engines with certain Honeywell part numbers (P/Ns) of Woodward fuel control unit (FCU) assemblies, installed. The AD number in the document headings is incorrect. Additionally, the Amendment number in the regulatory text is incorrect. This document corrects these two errors. In all other respects, the original document remains the same.

DATES: This final rule is effective on July 22, 2015. The effective date of AD 2015-12-04, Amendment 39-18177 (80 FR 34534, June 17, 2015) remains July 22, 2015.

ADDRESSES: You may examine the AD docket on the Internet at <http://www.regulations.gov/>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800-647-5527) is Document Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Joseph Costa, Aerospace Engineer, Los Angeles Aircraft Certification Office,

FAA, Transport Airplane Directorate, 3960 Paramount Blvd., Lakewood, CA 90712-4137; phone: 562-627-5246; fax: 562-627-5210; email: joseph.costa@faa.gov.

SUPPLEMENTARY INFORMATION: AD 2015-12-04, Amendment 39-18177, 80 FR 34534, June 17, 2015), requires initial and repetitive dimensional inspections of the affected fuel control drives and insertion of certain airplane operating procedures into the applicable flight manuals.

As published, the AD number in the document headings is incorrect. Additionally, the Amendment number in the regulatory text of AD 2015-12-04 is incorrect.

No other part of the final rule has been changed.

The effective date of AD 2015-12-04 remains July 22, 2015.

Correction of Non-Regulatory Text

In the **Federal Register** of June 17, 2015, AD 2015-12-04; Amendment 39-18177 (80 FR 34534) is corrected as follows:

On page 34534, in the 2nd column, on line 6, change “2014-12-04” to “2015-12-04”.

Correction of Regulatory Text

§ 39.13 [Corrected]

■ 2. The FAA amends § 39.13 by removing airworthiness directive (AD) 2006-15-08, Amendment 39-14688 (71 FR 41121, July 20, 2006), and adding the following new AD:

2015-12-04 Honeywell International Inc.:
Amendment 39-18177; Docket No. FAA-2006-23706; Directorate Identifier 2006-NE-03-AD.

(a) Effective Date

This AD is effective August 20, 2015.

(b) Affected ADs

This AD replaces AD 2006-15-08, Amendment 39-14688 (71 FR 41121, July 20, 2006).

(c) Applicability

This AD applies to all Honeywell International Inc. TPE331-1, -2, -2UA, -3U, -3UW, -5, -5A, -5AB, -5B, -6, -6A, -10, -10AV, -10GP, -10GT, -10P, -10R, -10T, -10U, -10UA, -10UF, -10UG, -10UGR, -10UR, -11U, -12JR, -12UA, -12UAR, and -12UHR turboprop engines with Honeywell part numbers (P/Ns) for Woodward fuel control unit (FCU) assemblies listed in Table 1 to paragraph (c) of this AD, installed.

TABLE 1 TO PARAGRAPH (C)—AFFECTED FCU ASSEMBLY P/Ns

| Group No. | Engine | FCU Assembly P/Ns |
|-----------|------------------------------|-----------------------------------------------------------------------------------------------|
| 1 | TPE331-1, -2, and -2UA | P/N 869199-13, -20, -21, -22, -23, -24, -25, -26, -27, -28, -29, -31, -32, -33, -34, and -35. |

TABLE 1 TO PARAGRAPH (C)—AFFECTED FCU ASSEMBLY P/Ns—Continued

| Group No. | Engine | FCU Assembly P/Ns |
|-----------|----------------------------------------------------------------------------------------------------|----------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------|
| 2* | TPE331-1, -2, and -2UA | P/N 869199-9, -10, -11, -12, -14, -16, -17, and -18. |
| 3 | TPE331-3U, -3UW, -5, -5A, -5AB, -5B, -6, -6A, -10AV, -10GP, -10GT, -10P, and -10T. | P/N 893561-7, -8, -9, -10, -11, -14, -15, -16, -20, -26, -27, and -29; or P/N 897770-1, -3, -7, -9, -10, -11, -12, -14, -15, -16, -25, -26, and -28. |
| 4* | TPE331-3U, -3UW, -5, -5B, -6, -6A, and -10T. | P/N 893561-4, -5, -12, and -13 or P/N 897770-5, -8, and -13. |
| 5 | TPE331-10, -10R, -10U, -10UA, -10UF, -10UG, -10UGR, -10UR, -11U, -12JR, -12UA, -12UAR, and -12UHR. | P/N 897375-2, -3, -4, -5, -8, -9, -10, -11, -12, -13, -14, -15, -16, -17, -19, -21, -24, -25, -26, and -27; or P/N 897780-1, -2, -3, -4, -5, -6, -7, -8, -9, -10, -11, -14, -15, -16, -17, -18, -19, -20, -21, -22, -23, -24, -25, -26, -27, -30, -32, -34, -36, -37, and -38; or P/N 893561-17, -18, and -19. |

* New/added FCU assembly P/Ns

(d) Unsafe Condition

We are issuing this AD to prevent failure of the fuel control drive that could result in damage to the engine and airplane.

(e) Compliance

Comply with this AD within the compliance times specified, unless already done.

(1) Inspection of Engines With FCU Assembly P/Ns in Groups 2 and 4

For FCU assembly P/Ns in Groups 2 and 4 listed in Table 1 to paragraph (c) of this AD:

(i) At the next scheduled inspection of the fuel control drive, or within 500 hours-in-

service (HIS) after the effective date of this AD, whichever occurs first, inspect the fuel control drive for wear.

(ii) Thereafter, re-inspect the fuel control drive within every 1,000 HIS since-last-inspection (SLI).

(2) Inspection of Engines With FCU Assembly P/Ns in Groups 1, 3, and 5

For FCU assembly P/Ns in Groups 1, 3, or 5 listed in Table 1 to paragraph (c) of this AD:

(i) If on the effective date of this AD the FCU assembly has 950 or more HIS SLI, inspect the fuel control drive for wear within 50 HIS from the effective date of this AD.

(ii) If on the effective date of this AD the FCU assembly has fewer than 950 HIS SLI, inspect the fuel control drive for wear before reaching 1,000 HIS.

(iii) Thereafter, re-inspect the fuel control drive for wear within every 1,000 HIS SLI.

(3) Airplane Operating Procedures

Within 60 days after the effective date of this AD, insert the information in Figure 1 to paragraph (e) of this AD, into the Emergency Procedures Section of the Airplane Flight Manual (AFM), Pilot Operating Handbook (POH), and the Manufacturer's Operating Manual (MOM).

Figure 1 to Paragraph (e) – Airplane Operating Procedures**NOTE**

Procedures in dotted line boxes are immediate action items to be performed by the pilot / flight crew.

**RAPID, UNCOMMANDED ACCELERATION DURING
ENGINE START (Propeller ON Start Locks)**

- Engine Start – Abort Immediately – Move condition lever to EMERGENCY STOP.

WARNING

Do not attempt to re-start engine. Report to maintenance.

ON GROUND or IN FLIGHT:

**RAPID, UNCOMMANDED INCREASE IN RPM, TORQUE,
FUEL FLOW AND/OR TURBINE TEMPERATURE
(Propeller OFF Start Locks)**

- Identify Malfunctioning Engine (multi-engine airplane) – Cross check for high torque, RPM, fuel flow, and turbine temperatures.
- Engine shut down - Move condition lever to EMERGENCY STOP.

WARNING

Never retard the power levers aft of flight idle in flight or on the ground.

WARNING

Do not attempt an engine re-start. Report to maintenance.

(f) Optional Terminating Action

Replacing the affected FCU assembly with an FAA-approved FCU assembly P/N not listed in this AD is terminating action for the initial and repetitive inspections required by this AD, and for inserting the information in Figure 1 to paragraph (e) of this AD into the AFM, POH, and MOM.

(g) Definitions

For the purposes of this AD:

(1) The “fuel control drive” is a series of mating splines located between the fuel pump and fuel control governor.

(2) The fuel control drive consists of four drive splines: the fuel pump internal spline, the fuel control external “quill shaft” spline, and the stub shaft internal and external splines.

(h) Alternative Methods of Compliance (AMOCs)

The Manager, Los Angeles Aircraft Certification Office, FAA, may approve

AMOCs for this AD. Use the procedures found in 14 CFR 39.19 to make your request.

(i) Related Information

(1) For more information about this AD, contact Joseph Costa, Aerospace Engineer, Los Angeles Aircraft Certification Office, FAA, Transport Airplane Directorate, 3960 Paramount Blvd., Lakewood, CA 90712–4137; phone: 562–627–5246; fax: 562–627–5210; email: joseph.costa@faa.gov.

(2) Information pertaining to operating recommendations for affected engines after a fuel control drive failure is contained in Honeywell International Inc., Operating Information Letter (OIL) OI331–12R6, dated May 26, 2009, for multi-engine airplanes; and in OIL OI331–18R4, dated May 26, 2009, for single-engine airplanes. Information on fuel control drive inspection can be found in Section 72–00–00 of the applicable TPE331 maintenance manuals. These Honeywell International Inc., OILs and the TPE331 maintenance manuals, which are not

incorporated by reference in this AD, can be obtained from Honeywell International Inc., using the contact information in paragraph (i)(3) of this AD.

(3) For service information identified in this AD, contact Honeywell International Inc., 111 S. 34th Street, Phoenix, AZ 85034–2802; Internet: <https://myaerospace.honeywell.com/wps/portal/lut>; phone: 800–601–3099.

(4) You may view this service information at the FAA, Engine & Propeller Directorate, 12 New England Executive Park, Burlington, MA. For information on the availability of this material at the FAA, call 781–238–7125.

(j) Material Incorporated by Reference

None.

Issued in Burlington, Massachusetts, on June 26, 2015.

Ann C. Mollica,

Acting Directorate Manager, Engine and Propeller Directorate, Aircraft Certification Service.

[FR Doc. 2015-16587 Filed 7-15-15; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2014-1123; Directorate Identifier 2014-CE-037-AD; Amendment [39-18209; AD 2015-06-02 R1]

RIN 2120-AA64

Airworthiness Directives; GA 8 Airvan (Pty) Ltd Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are revising an airworthiness directive (AD) 2015-06-02 for GA 8 Airvan (Pty) Ltd Model GA8-TC320 airplanes. This AD results from mandatory continuing airworthiness information (MCAI) issued by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as missing required engine mount fire seal washers, which could reduce the engine retention capability in the event of a fire. We are issuing this AD to require actions to address the unsafe condition on these products.

DATES: This AD is effective August 20, 2015.

The Director of the Federal Register approved the incorporation by reference of a certain other publication listed in this AD as of April 24, 2015 (80 FR 14810, March 20, 2015).

ADDRESSES: You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2014-1123; or in person at the Docket Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

For service information identified in this AD, contact GA 8 Airvan (Pty) Ltd, c/o GippsAero Pty Ltd, Attn: Technical Services, P.O. Box 881, Morwell Victoria 3840, Australia; telephone: + 61 03 5172 1200; fax: +61 03 5172 1201; email: techpubs@gippsaero.com;

Internet: <http://www.gippsaero.com/customer-support/technical-publications.aspx>. You may view this referenced service information at the FAA, Small Airplane Directorate, 901 Locust, Kansas City, Missouri 64106. For information on the availability of this material at the FAA, call (816) 329-4148. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2014-1123.

FOR FURTHER INFORMATION CONTACT:

Doug Rudolph, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4059; fax: (816) 329-4090; email: doug.rudolph@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to add an AD that would apply to GA 8 Airvan (Pty) Ltd Model GA8-TC320 airplanes. The NPRM was published in the **Federal Register** on April 17, 2015 (74 FR 21193), and proposed to revise AD 2015-06-02, Amendment 39-18120 (80 FR 14810; March 20, 2015).

The NPRM proposed to correct an unsafe condition for the specified products and was based on mandatory continuing airworthiness information originated by an airworthiness authority of another country. The MCAI states that:

A recent review of the engine mount installation on the GA8-TC 320 aircraft has highlighted the omission of engine mount fire seal washers during the assembly process.

The current engine mount configuration does not meet the certification basis for the aircraft, specifically regulation 23.865 of the Federal Aviation Regulations of the United States of America, where engine mounts located in designated fire zones are required to be suitably shielded so that they are capable of withstanding the effects of a fire.

The Gippsland Aeronautics GA8-TC 320 aircraft require the installation of an approved steel washer at each of the engine mount locations to address a potential risk of reduced engine retention capability in the event of a fire.

This AD, AD/GA8/8 Amdt 1, amends the applicability statement to be inclusive of the affected aircraft serial number range.

The MCAI can be found in the AD docket on the Internet at: <http://www.regulations.gov/#!documentDetail;D=FAA-2014-1123-0007>.

Comments

We gave the public the opportunity to participate in developing this AD. We

received no comments on the NPRM (74 FR 21193, April 17, 2015) or on the determination of the cost to the public.

Conclusion

We reviewed the relevant data and determined that air safety and the public interest require adopting the AD as proposed except for minor editorial changes. We have determined that these minor changes:

- Are consistent with the intent that was proposed in the NPRM (74 FR 21193, April 17, 2015) for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM (74 FR 21193, April 17, 2015).

Relative Service Information Under 14 CFR Part 51

We reviewed GippsAero Mandatory Service Bulletin SB-GA8-2014-115, Issue 1, dated October 6, 2014. The service bulletin describes procedures for inspecting the orientation of the engine isolator mounts to verify proper installation, re-installing if necessary, and installing steel washers on the forward side of each side of the engine isolator mounts. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section of this AD.

Costs of Compliance

We estimate that this AD will affect 13 products of U.S. registry. We also estimate that it would take about 5 work-hours per product to comply with the basic requirements of this AD. The average labor rate is \$85 per work-hour. Required parts would cost about \$10 per product.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition