

Actions	Compliance	Procedures
(1) Inspect the AFD-3010 adaptive flight display unit to determine the serial number.	Within the next 14 days after November 12, 2002 (the effective date of this AD), unless already accomplished.	In accordance with Rockwell Collins Service Bulletin 12, Revision No. 2 (AFD-3010-31-12), dated August 30, 2002.
(2) If the serial number of the AFD-3010 unit is listed as one of the affected units specified in Rockwell Collins Service Bulletin 12, Revision No. 2 (AFD-3010-31-12), dated August 30, 2002, then inspect the MFP386 ASIC device to determine the date code. (i) If the date code on the MFP386 ASIC device is 0128, return the unit to the manufacturer for modification and install the modified ADF-3010 unit. (ii) If the date code on the MFP386 ASIC is other than 0128, no manufacturer modification is necessary.	Prior to further flight after the inspection required in paragraph (d)(1) of this AD, unless already accomplished.	In accordance with Rockwell Collins Service Bulletin 12, Revision No. 2 (AFD-3010-31-12), dated August 30, 2002.
(3) Do not install, on any aircraft, an affected ADF-3010 adaptive flight display that contains a MFP386 ASIC device with a date code of 0128.	As of November 12, 2002 (the effective date of this AD).	Not applicable.

(e) *Can I comply with this AD in any other way?* You may use an alternative method of compliance or adjust the compliance time if:

(1) Your alternative method of compliance provides an equivalent level of safety; and

(2) The Manager, Wichita Aircraft Certification Office, approves your alternative. Submit your request through an FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Wichita Aircraft Certification Office.

Note: This AD applies to each airplane identified in paragraph (a) of this AD, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (e) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if you have not eliminated the unsafe condition, specific actions you propose to address it.

(f) *Where can I get information about any already-approved alternative methods of compliance?* Contact Roger A. Souter, FAA, Wichita Aircraft Certification Office (ACO), 1801 Airport Road, Room 100, Wichita, Kansas 67209; telephone: (316) 946-4134; facsimile: (316) 946-4407; e-mail: roger.souter@faa.gov.

(g) *What if I need to fly the airplane to another location to comply with this AD?* The FAA can issue a special flight permit under sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate your airplane to a location where you can accomplish the requirements of this AD.

(h) *Are any service bulletins incorporated into this AD by reference?* Actions required by this AD must be done in accordance with Rockwell Collins Service Bulletin 12 (AFD-3010-31-12), Revision No. 2, dated August

30, 2002. The Director of the Federal Register approved this incorporation by reference under 5 U.S.C. 552(a) and 1 CFR part 51. You can get copies from Rockwell Collins, Business and Regional Systems, 400 Collins Road Northeast, Cedar Rapids, Iowa 52498; telephone: (319) 295-1831. You may view this information at FAA, Central Region, Office of the Regional Counsel, 901 Locust, Room 506, Kansas City, Missouri, or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(i) *When does this amendment become effective?* This amendment becomes effective on November 12, 2002.

Issued in Kansas City, Missouri, on October 3, 2002.

James E. Jackson,

Acting Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 02-25717 Filed 10-15-02; 8:45 am]

BILLING CODE 4910-13-M

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2002-CE-41-AD; Amendment 39-12908; AD 2002-21-02]

RIN 2120-AA64

Airworthiness Directives; Cirrus Design Corporation Model SR20 and SR22 Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; request for comments.

SUMMARY: This amendment adopts a new airworthiness directive (AD) that applies to certain Cirrus Design

Corporation (Cirrus) Model SR20 and SR22 airplanes. This AD requires you to replace the self-locking retaining nut on the roll and yaw trim cartridges with a new self-locking retaining nut with a higher axial load capability. This AD is the result of a report that, during a production flight test, the self-locking retaining nut on the yaw trim cartridge came off. The actions specified by this AD are intended to prevent loss of the self-locking retaining nut on the roll and yaw trim cartridges during flight, which could result in jamming of the corresponding flight control system. Such jamming could lead to loss of control of the airplane.

DATES: This AD becomes effective on November 8, 2002.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in the regulation as of November 8, 2002.

The Federal Aviation Administration (FAA) must receive any comments on this rule on or before December 10, 2002.

ADDRESSES: Submit comments to FAA, Central Region, Office of the Regional Counsel, Attention: Rules Docket No. 2002-CE-41-AD, 901 Locust, Room 506, Kansas City, Missouri 64106. You may view any comments at this location between 8 a.m. and 4 p.m., Monday through Friday, except Federal holidays. You may also send comments electronically to the following address: 9-ACE-7-Docket@faa.gov. Comments sent electronically must contain "Docket No. 2002-CE-41-AD" in the subject line. If you send comments electronically as attached electronic files, the files must be formatted in

Microsoft Word 97 for Windows or ASCII text.

You may get the service information referenced in this AD from Cirrus Design Corporation, 4515 Taylor Circle, Duluth, MN 55811; telephone: (218) 727-2737; or electronically at the following address:

www.cirrusdesign.com/sb. You may view this information at FAA, Central Region, Office of the Regional Counsel, Attention: Rules Docket No. 2002-CE-41-AD, 901 Locust, Room 506, Kansas City, Missouri 64106; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT:

Gregory J. Michalik, Aerospace Engineer, FAA, Chicago Aircraft Certification Office, 2300 East Devon Avenue, Des Plaines, IL 60018; telephone: (847) 294-7135; facsimile: (847) 294-7834.

SUPPLEMENTARY INFORMATION:

Discussion

What Events Have Caused This AD?

The FAA received a report that, during a production flight test of one of the affected airplanes, the pilot attempted to apply right rudder but the control would not move. After investigation of this incident, we determined that the self-locking retaining nut on the shaft of the yaw trim cartridge came off and caused the yaw trim cartridge rod to jam when the right rudder was applied.

The yaw trim cartridge and the roll trim cartridge use the same shaft and self-locking retaining nut. The manufacturer has changed the design to include a new self-locking retaining nut with greater locking ability.

There have been no reports of this situation occurring on delivered airplanes in over 96,000 hours time-in-service on the fleet.

What Are the Consequences if the Condition Is Not Corrected?

This condition, if not corrected, could result in loss of the self-locking retaining nut on the roll and yaw trim cartridges. Such failure could lead to jamming of the corresponding flight control and cause loss of control of the airplane.

Is There Service Information That Applies to This Subject?

Cirrus Design Corporation has issued Alert Service Bulletin SB A20-27-06, Issued: September 20, 2002, and Alert Service Bulletin SB A22-27-03, Issued: September 20, 2002.

These service bulletins include procedures for installing a new self-

locking retaining nut on the roll trim and the yaw trim cartridges.

The FAA's Determination and an Explanation of the Provisions of This AD

What Has FAA Decided?

The FAA has reviewed all available information, including the service information referenced above; and determined that:

—The unsafe condition referenced in this document exists or could develop on other Cirrus Model SR20 and SR22 airplanes of the same type design;

—The actions specified in the previously-referenced service information (as specified in this AD) should be accomplished on the affected airplanes; and

—AD action should be taken in order to correct this unsafe condition.

What Does This AD Require?

This AD requires you to incorporate the actions in the previously-referenced service bulletin.

In preparation of this rule, we contacted type clubs and aircraft operators to obtain technical information and information on operational and economic impacts. We did not receive any information through these contacts. If received, we would have included, in the rulemaking docket, a discussion of any information that may have influenced this action.

Will I Have the Opportunity To Comment Prior to the Issuance of the Rule?

Because the unsafe condition described in this document could result in loss of flight controls, we find that notice and opportunity for public prior comment are impracticable. Therefore, good cause exists for making this amendment effective in less than 30 days.

Comments Invited

How Do I Comment on This AD?

Although this action is in the form of a final rule and was not preceded by notice and opportunity for public comment, FAA invites your comments on the rule. You may submit whatever written data, views, or arguments you choose. You need to include the rule's docket number and submit your comments to the address specified under the caption **ADDRESSES**. We will consider all comments received on or before the closing date specified above. We may amend this rule in light of comments received. Factual information that supports your ideas and suggestions is extremely helpful in evaluating the

effectiveness of the AD action and determining whether we need to take additional rulemaking action.

Are There Any Specific Portions of the AD I Should Pay Attention to?

We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of the rule that might suggest a need to modify the rule. You may view all comments we receive before and after the closing date of the rule in the Rules Docket. We will file a report in the Rules Docket that summarizes each FAA contact with the public that concerns the substantive parts of this AD.

How Can I Be Sure FAA Receives My Comment?

If you want us to acknowledge the receipt of your written comments, you must include a self-addressed, stamped postcard. On the postcard, write "Comments to Docket No. 2002-CE-41-AD." We will date stamp and mail the postcard back to you.

Regulatory Impact

Does This AD Impact Various Entities?

These regulations will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, FAA has determined that this final rule does not have federalism implications under Executive Order 13132.

Does This AD Involve a Significant Rule or Regulatory Action?

We have determined that this regulation is an emergency regulation that must be issued immediately to correct an unsafe condition in aircraft, and is not a significant regulatory action under Executive Order 12866. It has been determined further that this action involves an emergency regulation under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979). If it is determined that this emergency regulation otherwise would be significant under DOT Regulatory Policies and Procedures, a final regulatory evaluation will be prepared and placed in the Rules Docket (otherwise, an evaluation is not required). A copy of it, if filed, may be obtained from the Rules Docket.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. FAA amends § 39.13 by adding a new airworthiness directive (AD) to read as follows:

2002–21–02 Cirrus Design Corporation:
Amendment 39–12908; Docket No. 2002–CE–41–AD.

(a) *What airplanes are affected by this AD?*
This AD applies to the following airplane models and serial numbers that are certificated in any category:

Model	Serial numbers
SR20	1005 through 1241, except 1235, 1237, and 1238.
SR22	0002 through 0333, except 0309, 0322, 0323, and 0328.

(b) *Who must comply with this AD?*

Anyone who wishes to operate any of the airplanes identified in paragraph (a) of this AD must comply with this AD.

(c) *What problem does this AD address?*

The actions specified by this AD are intended to prevent loss of the self-locking retaining nut on the roll and yaw trim cartridges during flight, which could result in jamming of the corresponding flight control system. Such jamming could lead to loss of control of the airplane.

(d) *What must I do to address this problem?* To address this problem, you must accomplish the following actions:

Actions	Compliance	Procedures
(1) Replace the self-locking retaining nut on the yaw trim cartridge and the roll trim cartridge with a new self-locking retaining nut, part number MS21044N3.	Within the next 10 hours time-in-service after November 8, 2002 (the effective date of this AD), unless already accomplished.	In accordance with Cirrus Alert Service Bulletin SB A20–27–06, Issued: September 20, 2002, and Cirrus Alert Service Bulletin SB A22–27–03, Issued: September 20, 2002, as applicable.
(2) Do not install any self-locking retaining nut on the yaw trim cartridge or the roll trim cartridge that is not part number MS21044N3.	As of November 8, 2002 (the effective date of this AD).	Not applicable.

(e) *Can I comply with this AD in any other way?* You may use an alternative method of compliance or adjust the compliance time if:

- (1) Your alternative method of compliance provides an equivalent level of safety; and
- (2) The Manager, Chicago Aircraft Certification Office (ACO), approves your alternative. Submit your request through an FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Chicago ACO.

Note: This AD applies to each airplane identified in paragraph (a) of this AD, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (e) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if you have not eliminated the unsafe condition, specific actions you propose to address it.

(f) *Where can I get information about any already-approved alternative methods of compliance?* Contact Gregory J. Michalik, Aerospace Engineer, FAA, Chicago ACO, 2300 East Devon Avenue, Des Plaines, IL 60018; telephone: (847) 294–7135; facsimile: (847) 294–7834.

(g) *What if I need to fly the airplane to another location to comply with this AD?* The FAA can issue a special flight permit under sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate your airplane to a location

where you can accomplish the requirements of this AD.

(h) *Are any service bulletins incorporated into this AD by reference?* Actions required by this AD must be done in accordance with in Cirrus Alert Service Bulletin SB A20–27–06, Issued: September 20, 2002, and Cirrus Alert Service Bulletin SB A22–27–03, Issued: September 20, 2002. The Director of the Federal Register approved this incorporation by reference under 5 U.S.C. 552(a) and 1 CFR part 51. You can get copies from Cirrus Design Corporation, 4515 Taylor Circle, Duluth, MN 55811; telephone: (218) 727–2737; or electronically at the following address: www.cirrusdesign.com/sb. You may view this information at FAA, Central Region, Office of the Regional Counsel, 901 Locust, Room 506, Kansas City, Missouri, or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(i) *When does this amendment become effective?* This amendment becomes effective on November 8, 2002.

Issued in Kansas City, Missouri, on October 7, 2002.

Michael Gallagher,

Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 02–26052 Filed 10–15–02; 8:45 am]

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DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. 2002–SW–46–AD; Amendment 39–12910; AD 2002–21–04]

RIN 2120–AA64

Airworthiness Directives; Agusta S.p.A. Model A119 Helicopters

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; request for comments.

SUMMARY: This amendment supersedes an existing emergency airworthiness directive (EAD) for Agusta S.p.A. (Agusta) Model A119 helicopters. EAD 2002–17–52, which was issued on August 21, 2002, was sent to all known U.S. owners and operators of Agusta Model A119 helicopters by individual letters. That AD currently requires installing a placard in the helicopter and marking the airspeed indication at 132 knots indicated airspeed (KIAS) before further flight; visually checking the tail rotor blades on both sides for a crack before each start of the helicopter engine; visually inspecting the tail rotor blades with a 5x or higher magnifying glass and conducting a dye-penetrant inspection if you are unable to determine by the visual inspection