

repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent failure of the attachment bolts of the wing-to-fuselage attachment fittings due to stress corrosion cracking, which could result in reduced structural integrity of the airplane, accomplish the following:

(a) Within 12 months after the effective date of this AD, accomplish the requirements of paragraphs (a)(1), (a)(2), and (a)(3) of this AD, in accordance with Bombardier Service Bulletin S.B. 7-57-37, dated August 8, 1997.

(1) Remove all attachment bolts (one at a time), and pre-load indicating (PLI) washers of the wing-to-fuselage attachment fittings.

(2) Perform a one-time visual inspection to detect corrosion of each attachment bolt. If any corrosion is detected, within 10 days after accomplishing the visual inspection, or within 10 days after the effective date of this AD, whichever occurs later, submit a report of the inspection results to Bombardier, Inc., Bombardier Regional Aircraft Division, Garratt Boulevard, Downsview, Ontario M3K 1Y5, Canada. Information collection requirements contained in this regulation have been approved by the Office of Management and Budget (OMB) under the provisions of the Paperwork Reduction Act of 1980 (44 U.S.C. 3501 *et seq.*) and have been assigned OMB Control Number 2120-0056.

(3) Install new attachment bolts (one at a time), and new PLI washers of the wing-to-fuselage attachment fittings.

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, New York Aircraft Certification Office (ACO), FAA, Engine and Propeller Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, New York ACO.

**Note 2:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the New York ACO.

(c) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

**Note 3:** The subject of this AD is addressed in Canadian airworthiness directive CF-98-12, dated June 24, 1998.

Issued in Renton, Washington, on November 20, 1998.

**Darrell M. Pederson,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

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## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 98-NM-289-AD]

RIN 2120-AA64

#### Airworthiness Directives; Short Brothers Model SD3-60 SHERPA Series Airplanes

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** This document proposes the adoption of a new airworthiness directive (AD) that is applicable to all Short Brothers Model SD3-60 SHERPA series airplanes. This proposal would require a one-time visual inspection to determine the part number of the power control cable assemblies and pulleys of the engine controls; and replacement of the power control cable assemblies and pulleys (as applicable) with new parts, if necessary. This proposal is prompted by issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by the proposed AD are intended to prevent breakage of the power control cable assemblies due to the inflexible construction of the cable, which could result in loss of engine power and consequent reduced controllability of the airplane.

**DATES:** Comments must be received by December 30, 1998.

**ADDRESSES:** Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 98-NM-289-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056. Comments may be inspected at this location between 9:00 a.m. and 3:00 p.m., Monday through Friday, except Federal holidays.

The service information referenced in the proposed rule may be obtained from Short Brothers, Airworthiness & Engineering Quality, P.O. Box 241, Airport Road, Belfast BT3 9DZ, Northern Ireland. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington.

**FOR FURTHER INFORMATION CONTACT:** Norman B. Martenson, Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2110; fax (425) 227-1149.

## SUPPLEMENTARY INFORMATION:

### Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this notice may be changed in light of the comments received.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this notice must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 98-NM-289-AD." The postcard will be date stamped and returned to the commenter.

### Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 98-NM-289-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056.

### Discussion

The Civil Aviation Authority (CAA), which is the airworthiness authority for the United Kingdom, notified the FAA that an unsafe condition may exist on all Short Brothers Model SD3-60 SHERPA series airplanes. The CAA advises that the existing power control cable assemblies are subject to breakage, which is caused by the inflexible construction of the cable. The manufacturer has introduced a more flexible construction of the cable, which was incorporated during production of the subject airplanes, however, the possibility does exist that not all airplanes were modified. Breakage of the power control cable assemblies due to the inflexible construction of the cable, if not corrected, could result in loss of engine power and consequent reduced controllability of the airplane.

### Explanation of Relevant Service Information

The manufacturer has issued Service Bulletin SD3-60 SHERPA-76-1, dated July 1998, which describes procedures for a one-time visual inspection to determine the part number of the power control cable assemblies and pulleys of the engine controls; and replacement of the power control cable assemblies and pulleys (as applicable) with new parts, if necessary. Accomplishment of the actions specified in the service bulletin is intended to adequately address the identified unsafe condition. The CAA classified this service bulletin as mandatory and issued British airworthiness directive 004-07-98 in order to assure the continued airworthiness of these airplanes in the United Kingdom.

### FAA's Conclusions

This airplane model is manufactured in the United Kingdom and is type certificated for operation in the United States under the provisions of section 21.29 of the Federal Aviation Regulations (14 CFR 21.29) and the applicable bilateral airworthiness agreement. Pursuant to this bilateral airworthiness agreement, the CAA has kept the FAA informed of the situation described above. The FAA has examined the findings of the CAA, reviewed all available information, and determined that AD action is necessary for products of this type design that are certificated for operation in the United States.

### Explanation of Requirements of Proposed Rule

Since an unsafe condition has been identified that is likely to exist or develop on other airplanes of the same type design registered in the United States, the proposed AD would require accomplishment of the actions specified in the service bulletin described previously.

### Cost Impact

The FAA estimates that 28 airplanes of U.S. registry would be affected by this proposed AD, that it would take approximately 15 work hours per airplane to accomplish the proposed inspection, and that the average labor rate is \$60 per work hour. Based on these figures, the cost impact of the proposed inspection by this AD on U.S. operators is estimated to be \$25,200, or \$900 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the proposed requirements of this AD action, and that no operator would

accomplish those actions in the future if this AD were not adopted.

### Regulatory Impact

The regulations proposed herein would not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this proposal would not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption ADDRESSES.

### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

### The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

### PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

**Short Brothers, PLC:** Docket 98-NM-289-AD.

**Applicability:** All Model SD3-60 SHERPA series airplanes, certificated in any category.

**Note 1:** This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the

owner/operator must request approval for an alternative method of compliance in accordance with paragraph (c) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent breakage of the power control cable assemblies due to the inflexible construction of the cable, which could result in loss of engine power and consequent reduced controllability of the airplane, accomplish the following:

(a) Within 1,200 flight hours after the effective date of this AD, perform a one-time visual inspection to determine the part number (P/N) of the power control cable assemblies and pulleys of the engine controls, in accordance with Part A of the Accomplishment Instructions of Shorts Service Bulletin SD3-60 SHERPA-76-1, dated July 1998.

(1) If any power control cable assembly having P/N SD3-47-1091 or SD3-47-1094 is found, prior to further flight, replace the power control cable assembly with a new power control cable assembly in accordance with Part B of the Accomplishment Instructions of the service bulletin.

(2) If any pulley having P/N C181605 is found, prior to further flight, replace the pulley with a new pulley in accordance with Part C of the Accomplishment Instructions of the service bulletin.

(b) As of the effective date of this AD, no person shall install on the engine controls of any airplane a cable assembly having P/N SD3-47-1091 or SD3-47-1094, or any pulley having P/N C181605.

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

**Note 2:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

(d) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

**Note 3:** The subject of this AD is addressed in British airworthiness directive 004-07-98.

Issued in Renton, Washington, on November 20, 1998.

**Darrell M. Pederson,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

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