

2006-16-13 PILATUS AIRCRAFT LTD:
Amendment 39-14713; Docket No.
FAA-2006-24954; Directorate Identifier
2006-CE-30-AD.

Effective Date

(a) This airworthiness directive (AD) becomes effective September 18, 2006.

Affected ADs

(b) None.

Applicability

(c) This AD applies to Models PC-12 and PC-12/45 airplanes; manufacturer serial numbers 101 through 617 inclusive, certificated in any U.S. category.

Reason

(d) The mandatory continuing airworthiness information (MCAI) states that the aircraft manufacturer has identified drill damage on some Frame 21 (FR21) lug fittings on the production line and during a number of midlife wing lug inspections. It is thought that the damage found on the FR21 lug fittings occurred during assembly of the airplane. Depending on the size and location of the possible damage, if not corrected, the fatigue life of the wing attachment lugs on FR21 may be affected. The MCAI requires a one-time inspection of the FR21 adjacent to the wing upper-attachment lugs, left and right, and a repair if necessary.

Actions and Compliance

(e) Unless already done, do the following except as stated in paragraph (f) below.

(1) Within the next 100 hours time-in-service (TIS) after September 15, 2006 (the effective date of this AD), perform an inspection of FR21 in the area of the outer sidewall frame attachment lug forward and aft side faces, left and right, to determine if there is any damage that may have been made with a drill. Follow Pilatus Aircraft Ltd. PC12 Service Bulletin No. 53-004, dated February 10, 2006.

(2) Within the next 100 hours TIS after September 18, 2006 (the effective date of this AD), perform an inspection of FR21 in the area of the top surface of the wing upper-attachment lugs, left and right, to determine if there is any damage that may have been made with a drill. Follow Pilatus Aircraft Ltd. PC12 Service Bulletin No. 53-004, dated February 10, 2006.

(3) If during the inspection required by paragraph (e)(1) of this AD any damage less than 0.1 mm (0.0040 inch) on any FR21 is found, prior to further flight, repair the damaged FR21 in accordance with Pilatus Aircraft Ltd. PC12 Service Bulletin No. 53-004, dated February 10, 2006.

(4) If during the inspection required in paragraph (e)(1) of this AD any damage equal to or greater than 0.1 mm (0.0040 inch) on any FR21 is found, prior to further flight contact Pilatus Aircraft Ltd. for an FAA-approved repair solution and incorporate the repair.

(5) If during the inspection required by paragraph (e)(2) of this AD any damage less than 1 mm (0.040 inch) depth on any FR21 wing attachment lug top surface is found, prior to further flight, repair the damaged FR21 in accordance with Pilatus Aircraft Ltd.

PC12 Service Bulletin No. 53-004, dated February 10, 2006.

(6) If during the inspection required by paragraph (e)(2) of this AD any damage equal to or greater than 1 mm (0.040 inch) depth on any FR21 wing attachment lug top surface is found, prior to further flight contact Pilatus Aircraft Ltd. for an FAA-approved repair solution and incorporate the repair.

FAA AD Differences

(f) None.

Other FAA AD Provisions

(g) The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs):* The Manager, Standards Staff, FAA, ATTN: Doug Rudolph, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri, 64106; telephone: (816) 329-4059; facsimile: (816) 329-4090, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19.

(2) *Return to Airworthiness:* When complying with this AD, perform FAA-approved corrective actions before returning the product to an airworthy condition.

(3) *Reporting Requirements:* For any reporting requirement in this AD, under the provisions of the Paperwork Reduction Act, the Office of Management and Budget (OMB) has approved the information collection requirements and has assigned OMB Control Number 2120-0056.

Related Information

(h) This AD is related to Federal Office for Civil Aviation AD HB-2006-223, effective date April 20, 2006, which references Pilatus Aircraft Ltd. PC12 Service Bulletin No. 53-004, dated February 10, 2006.

Material Incorporated by Reference

(i) You must use Pilatus Aircraft Ltd. PC12 Service Bulletin No. 53-004, dated February 10, 2006, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact the Pilatus Aircraft Ltd., Customer Support Manager, CH-6371 STANS, Switzerland; telephone: 41 41 619 6208; facsimile: +41 41 619 7311; email: SupportPC12@pilatus-aircraft.com.

(3) You may review copies at the FAA, Central Region, Office of the Regional Counsel, 901 Locust, Room 506, Kansas City, Missouri 64106; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Kansas City, Missouri, on August 4, 2006.

John R. Colomy,

Acting Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. E6-13016 Filed 8-11-06; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2005-21242; Directorate Identifier 2005-NE-09-AD; Amendment 39-14721; AD 2006-02-08R1]

RIN 2120-AA64

Airworthiness Directives; Turbomeca Arriel 1B, 1D, 1D1, and 1S1 Turboshaft Engines

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: The FAA is revising an existing airworthiness directive (AD) for certain Turbomeca Arriel 1B, 1D, 1D1, and 1S1 turboshaft engines. That AD currently requires initial and repetitive position checks of the gas generator 2nd stage turbine blades on all Turbomeca Arriel 1B, 1D, 1D1, and 1S1 turboshaft engines. That AD also currently requires initial and repetitive replacements of 2nd stage turbines on 1B, 1D, and 1D1 engines only. This AD revision requires the same actions, but would relax the compliance times for initially replacing 2nd stage turbines in Arriel 1B, 1D, and 1D1 turboshaft engines. We are issuing this AD revision to clarify and relax the AD compliance times for 2nd stage turbine initial replacement on Arriel 1B, 1D, and 1D1 turboshaft engines. We are also issuing this AD revision to prevent in-flight engine shutdown and subsequent forced autorotation landing or accident.

DATES: This AD becomes effective September 13, 2006. The Director of the Federal Register previously approved the incorporation by reference of certain publications listed in the regulations as of February 28, 2006 (71 FR 3754, January 24, 2006).

ADDRESSES: You can get the service information identified in this AD from Turbomeca, 40220 Tarnos, France; telephone +33 05 59 74 40 00, fax +33 05 59 74 45 15.

You may examine the AD docket on the Internet at <http://dms.dot.gov> or in Room PL-401 on the plaza level of the Nassif Building, 400 Seventh Street, SW., Washington, DC.

FOR FURTHER INFORMATION CONTACT: Christopher Spinney, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803; telephone (781) 238-7175, fax (781) 238-7199.

SUPPLEMENTARY INFORMATION: The FAA proposed to amend 14 CFR part 39 with a proposed AD. The proposed AD revision applies to certain Turbomeca Arriel 1B, 1D, 1D1, and 1S1 turboshaft engines. We published the proposed AD revision in the **Federal Register** on April 17, 2006 (71 FR 3754). That action proposed to require initial and repetitive position checks of the gas generator 2nd stage turbine blades on all Turbomeca Arriel 1B, 1D, 1D1, and 1S1 turboshaft engines. That action also proposed to require initial and repetitive replacements of 2nd stage turbines on 1B, 1D, and 1D1 engines only, but proposed to relax the compliance times for initially replacing 2nd stage turbines in Arriel 1B, 1D, and 1D1 turboshaft engines.

Examining the AD Docket

You may examine the docket that contains the AD, any comments received, and any final disposition in person at the Docket Management Facility Docket Office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The Docket Office (telephone (800) 647-5227) is located on the plaza level of the Department of Transportation Nassif Building at the street address stated in **ADDRESSES**. Comments will be available in the AD docket shortly after the DMS receives them.

Comments

We provided the public the opportunity to participate in the development of this AD. We received no comments on the proposal or on the determination of the cost to the public.

Conclusion

We have carefully reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed.

Costs of Compliance

We estimate that this AD revision would affect 721 engines installed on helicopters of U.S. registry. We also estimate that it will take about 2 work-hours per engine to inspect all 721 engines and 40 work-hours per engine to replace about 571 2nd stage turbines on 1B and 1D1 engines, and that the average labor rate is \$80 per work-hour. Required parts will cost about \$3,200 per engine. Based on these figures, we estimate the total cost of the AD revision to U.S. operators to be \$4,249,760.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue

rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866;
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
- (3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a summary of the costs to comply with this AD and placed it in the AD Docket. You may get a copy of this summary at the address listed under **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

■ Accordingly, under the authority delegated to me by the Administrator, the Federal Aviation Administration amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. The FAA amends § 39.13 by removing Amendment 39-14460 (71 FR 3754, January 24, 2006), and by adding a new airworthiness directive, Amendment 39-14721, to read as follows:

2006-02-08R1 Turbomeca: Amendment 39-14721. Docket No. FAA-2005-21242; Directorate Identifier 2005-NE-09-AD.

Effective Date

(a) This airworthiness directive (AD) becomes effective September 13, 2006.

Affected ADs

(b) This AD revises AD 2006-02-08, Amendment 39-14460.

Applicability

(c) This AD revision applies to Turbomeca Arriel 1B engines fitted with 2nd stage turbine modification TU 148, and Arriel 1D, 1D1, and 1S1 engines. Arriel 1B engines are installed on, but not limited to, Eurocopter France AS-350B and AS-350A "Ecureuil" helicopters. Arriel 1D engines are installed on, but not limited to, Eurocopter France AS-350B1 "Ecureuil" helicopters. Arriel 1D1 engines are installed on, but not limited to, Eurocopter France AS-350B2 "Ecureuil" helicopters. Arriel 1S1 engines are installed on, but not limited to, Sikorsky Aircraft S-76A and S-76C helicopters.

Unsafe Condition

(d) This AD revision results from a request by Turbomeca to clarify the compliance times for 2nd stage turbine initial replacement on Arriel 1B, 1D, and 1D1 turboshaft engines. We are issuing this AD revision to clarify and relax the AD compliance times for 2nd stage turbine initial replacement on Arriel 1B, 1D, and 1D1 turboshaft engines. We are also issuing this AD revision to prevent in-flight engine shutdown and subsequent forced autorotation landing or accident.

Compliance

(e) You are responsible for having the actions required by this AD revision performed within the compliance times specified unless the actions have already been done.

Initial Relative Position Check of 2nd Stage Turbine Blades

(f) Do an initial relative position check of the 2nd stage turbine blades using the Turbomeca mandatory alert service bulletins (ASBs) specified in the following Table 1. Do the check before reaching any of the intervals specified in Table 1 or within 50 hours time-in-service after the effective date of this AD, whichever occurs later.

TABLE 1.—INITIAL AND REPETITIVE RELATIVE POSITION CHECK INTERVALS OF 2ND STAGE TURBINE BLADE

Turbomeca engine model	Initial relative position check interval	Repetitive interval	Mandatory alert service bulletin
Arriel 1B (modified per TU 148).	Within 1,200 hours time-since-new (TSN) or time-since-overhaul (TSO) or 3,500 cycles-since-new (CSN) or cycles-since-overhaul (CSO), whichever occurs earlier.	Within 200 hours time-in-service-since-last-relative-position-check (TSLRPC).	A292 72 0807, dated March 24, 2004.
Arriel 1D1 and Arriel 1D	Within 1,200 hours TSN or TSO or 3,500 hours CSN or CSO, whichever occurs earlier.	Within 150 hours TSLRPC.	A292 72 0809, Update No. 1, dated October 4, 2005.
Arriel 1S1	Within 1,200 hours TSN or TSO or 3,500 hours CSN or CSO, whichever occurs earlier.	Within 150 hours TSLRPC	A292 72 0810, dated March 24, 2004.

Repetitive Relative Position Check of 2nd Stage Turbine Blades

(g) Recheck the relative position of 2nd stage turbine blades at the TSLRPC intervals specified in Table 1 of this AD, using the mandatory ASBs indicated.

Credit for Previous Relative Position Checks

(h) Relative position checks of 2nd stage turbine blades done using Turbomeca Service Bulletin A292 72 0263, Update 1, 2, 3, or 4, may be used to show compliance with the initial requirements of paragraph (f) of this AD.

Initial Replacement of 2nd Stage Turbines on Arriel 1B, 1D, and 1D1 Engines

(i) Initially replace the 2nd stage turbine with a new or overhauled 2nd stage turbine as follows:

(1) On or before August 31, 2006, replace the 2nd stage turbine with a new or overhauled 2nd stage turbine:

(i) As soon as practicable after accumulating 1,500 hours TSN or TSO for Arriel 1D and 1D1 engines.

(ii) As soon as practicable after accumulating 2,200 hours TSN or TSO for Arriel 1B engines.

(2) After August 31, 2006, replace the 2nd stage turbine with a new or overhauled 2nd stage turbine:

(i) Before accumulating 1,500 hours TSN or TSO for Arriel 1D and 1D1 engines.

(ii) Before accumulating 2,200 hours TSN or TSO for Arriel 1B engines.

Repetitive Replacements of 2nd Stage Turbines on Arriel 1B, 1D, and 1D1 Engines

(j) Thereafter, replace the 2nd stage turbine with a new or overhauled 2nd stage turbine within every 1,500 hours TSN or TSO for Arriel 1D and 1D1 engines, and within every 2,200 hours TSN or TSO for Arriel 1B engines.

Criteria for Overhauled 2nd Stage Turbines

(k) Do the following to overhauled 2nd stage turbines, referenced in paragraphs (i) and (j) of this AD:

(1) You must install new blades in the 2nd stage turbines of overhauled Arriel 1D and 1D1 engines.

(2) You may install either overhauled or new blades in the 2nd stage turbines of overhauled Arriel 1B engines.

Relative Position Check Continuing Compliance Requirements

(l) All 2nd stage turbines, including those that are new or overhauled, must continue to comply with relative position check requirements of paragraphs (f) and (j) of this AD.

Alternative Methods of Compliance

(m) The Manager, Engine Certification Office, has the authority to approve

alternative methods of compliance for this AD if requested using the procedures found in 14 CFR 39.19.

Related Information

(n) DGAC airworthiness directive F-2004-047 R1, dated October 26, 2005, also addresses the subject of this AD.

Material Incorporated by Reference

(o) You must use the service information specified in Table 2 of this AD to perform the actions required by this AD. The Director of the Federal Register previously approved the incorporation by reference of the documents listed in Table 2 of this AD in accordance with 5 U.S.C. 552(a) and 1 CFR part 51, as of February 28, 2006 (71 FR 3754, January 24, 2006). Contact Turbomeca, 40220 Tarnos, France; telephone +33 05 59 74 40 00, fax +33 05 59 74 45 15, for a copy of this service information. You may review copies at the Docket Management Facility; U.S. Department of Transportation, 400 Seventh Street, SW., Nassif Building, Room PL-401, Washington, DC 20590-0001, on the Internet at <http://dms.dot.gov>, or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

TABLE 2.—INCORPORATION BY REFERENCE

Turbomeca mandatory alert service bulletin no.	Page	Update number	Date
A292 72 0807	ALL	Original	March 24, 2004.
Total Pages: 17			
A292 72 0809	ALL	1	October 4, 2005.
Total Pages: 18			
A292 72 0810	ALL	Original	March 24, 2004.
Total Pages: 14			

Issued in Burlington, Massachusetts, on August 8, 2006.
Francis A. Favara,
Manager, Engine and Propeller Directorate, Aircraft Certification Service.
 [FR Doc. E6-13249 Filed 8-11-06; 8:45 am]
BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION
Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2006-24255; Directorate Identifier 2006-CE-25-AD; Amendment 39-14720; AD 2006-16-20]

RIN 2120-AA64

Airworthiness Directives; DG Flugzeugbau GmbH Model DG-1000S Sailplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for all DG Flugzeugbau GmbH Model DG-1000S sailplanes. This AD requires you to modify the elevator control at the stabilizer assembly, replace a placard on the fin, and incorporate changes in the FAA-approved sailplane flight manual (SFM). This AD results from mandatory continuing airworthiness information (MCAI) issued by the airworthiness authority for Germany. We are issuing this AD to prevent the rigging of the

horizontal stabilizer without properly connecting the elevator, which, if not prevented, could lead to an inoperative elevator. An inoperative elevator could lead to loss of control of the sailplane.

DATES: This AD becomes effective on September 18, 2006.

As of September 18, 2006, the Director of the Federal Register approved the incorporation by reference of certain publications listed in the regulation.

ADDRESSES: For service information identified in this AD, contact DG-Flugzeugbau, Postbox 41 20, D-76625 Bruchsal, Federal Republic of Germany; telephone: ++49 7257 890; facsimile: ++45 7257 8922; e-mail: <http://www.dg-flugzeugbau.de>.

To view the AD docket, go to the Docket Management Facility; U.S. Department of Transportation, 400 Seventh Street, SW., Nassif Building, Room PL-401, Washington, DC 20590-001 or on the Internet at <http://dms.dot.gov>. The docket number is FAA-2006-24255; Directorate Identifier 2006-CE-25-AD.

FOR FURTHER INFORMATION CONTACT: Gregory Davison, Glider Project Manager, ACE-112, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4130; facsimile: (816) 329-4090.

SUPPLEMENTARY INFORMATION:

Discussion

On May 9, 2006, we issued a proposal to amend part 39 of the Federal Aviation

Regulations (14 CFR part 39) to include an AD that would apply to all DG Flugzeugbau GmbH Model DG-1000S sailplanes. This proposal was published in the **Federal Register** as a notice of proposed rulemaking (NPRM) on May 16, 2006 (71 FR 28287). The NPRM proposed to require you to modify the elevator control at the stabilizer assembly, replace a placard on the fin, and incorporate changes in the FAA-approved SFM.

Comments

We provided the public the opportunity to participate in developing this AD. We received no comments on the proposal or on the determination of the cost to the public.

Conclusion

We have carefully reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed except for minor editorial corrections. We have determined that these minor corrections:

- Are consistent with the intent that was proposed in the NPRM for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM.

Costs of Compliance

We estimate that this AD affects 8 sailplanes in the U.S. registry.

We estimate the following costs to do the modification and replacement of the placard on the fin:

Labor cost	Parts cost	Total cost per sailplane	Total cost on U.S. operators
2 work-hours × \$80 per hour = \$160	\$60	\$220	8 × \$220 = \$1,760

We estimate the following costs to do the incorporation of changes in the FAA-approved SFM:

Labor cost	Parts cost	Total cost per sailplane	Total cost on U.S. operators
1 work-hour × \$80 per hour = \$80	Not applicable	\$80	8 × \$80 = \$640

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, Section 106 describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation

is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this AD.

Regulatory Findings

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between