

New Requirements of This AD*Repetitive Inspections*

(k) At the later of the applicable times specified in the "Threshold (FC)" and "Grace Period" columns of Tables 1 and 2 in paragraph 1.E of the applicable service bulletin specified in Table 2 of this AD: Do

an ultrasonic inspection or HFEC inspection, including rework of the pressure diaphragm, for cracks in the lower flanges of the left and right gantries 1 through 5 inclusive between FR47 and FR54, in accordance with the Accomplishment Instructions of the applicable service bulletin in Table 2 of this

AD. Repeat the inspection at the applicable times specified in the "Interval (FC)" column of Tables 1 and 2 in paragraph 1.E of the applicable service bulletin in Table 2 of this AD. Accomplishment of the initial inspection ends the inspections required by paragraphs (f), (h), and (i) of this AD.

TABLE 2.—SERVICE BULLETINS

Airbus service bulletin—	For airplanes identified in—
(1) A300–53–0379, Revision 01, dated October 4, 2005	Paragraphs (c)(1) and (c)(2) of this AD.
(2) A300–53–6152, Revision 01, dated October 4, 2005	Paragraphs (c)(3) through (c)(6) of this AD inclusive.

Corrective Action

(l) If any crack is detected during any ultrasonic or HFEC inspection required by paragraph (k) of this AD, before further flight, repair the crack in accordance with the

Accomplishment Instructions of the applicable service bulletin in Table 2 of this AD, except as provided by paragraph (n) of this AD.

Optional Terminating Actions

(m) Accomplishment of the actions specified in Table 3 of this AD ends the repetitive inspections required by paragraph (k) of this AD.

TABLE 3.—OPTIONAL TERMINATING ACTIONS

Before or at the same time with—	Reinforce—	By doing all the actions in accordance with the Accomplishment Instructions of Airbus Service Bulletin—	For airplanes identified in—
(1) The actions required by paragraph (k) of this AD and the action specified in paragraph (m)(2) of this AD.	The flanges of the left and right portals 1 through 5 inclusive between FR47 and FR54 of the landing gear, including a rotating probe inspection for cracks of holes and repair if necessary.	A300–53–0380, dated August 5, 2005, except as provided by paragraph (n) of this AD.	Paragraphs (c)(1) and (c)(2) of this AD.
(2) The actions required by paragraph (k) of this AD.	Portals 3, 4, and 5 of the plates/skin.	A300–53–6153, dated August 24, 2005, except as provided by paragraph (n) of this AD.	Paragraphs (c)(3) through (c)(6) of this AD inclusive.
		A300–53–0360, dated May 3, 2002, except as provided by paragraph (n) of this AD.	Paragraphs (c)(1) and (c)(2) of this AD
		A300–53–6132, dated February 5, 2002, except as provided by paragraph (n) of this AD.	Paragraphs (c)(3) through (c)(6) of this AD inclusive

Repair of Certain Cracks

(n) Where the applicable service bulletin recommends contacting Airbus for appropriate action: Before further flight, repair the crack in accordance with a method approved by the Manager, International Branch, ANM–116; or the DGAC (or its delegated agent).

accordance with the procedures found in 14 CFR 39.19.

(2) Before using any AMOC approved in accordance with § 39.19 on any airplane to which the AMOC applies, notify the appropriate principal inspector in the FAA Flight Standards Certificate Holding District Office.

Credit for Original Service Bulletins

(o) Accomplishing the inspections and repair before the effective date of this AD in accordance with Airbus Service Bulletin A300–53–0379, dated May 9, 2005, or Airbus Service Bulletin A300–53–6152, dated May 9, 2005, as applicable, is acceptable for compliance with the corresponding requirements of paragraphs (k) and (l) of this AD.

Related Information

(r) French airworthiness directive F–2005–091 R1, issued September 28, 2005, also addresses the subject of this AD.

Issued in Renton, Washington, on May 8, 2006.

Ali Bahrami,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. E6–7477 Filed 5–16–06; 8:45 am]

BILLING CODE 4910–13–P

No Inspection Report

(p) Although the service bulletins in this AD specify to submit certain information to the manufacturer, this AD does not include that requirement.

Alternative Methods of Compliance (AMOCs)

(q)(1) The Manager, International Branch, ANM–116, has the authority to approve AMOCs for this AD, if requested in

DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA–2006–24780; Directorate Identifier 2006–NM–069–AD]

RIN 2120–AA64

Airworthiness Directives; McDonnell Douglas Model DC–10–10, DC–10–10F, DC–10–15, DC–10–30, DC–10–30F (KC–10A and KDC–10), DC–10–40, and DC–10–40F Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: The FAA proposes to adopt a new airworthiness directive (AD) for certain McDonnell Douglas airplanes, identified above. This proposed AD would require installing or replacing with improved parts, as applicable, the bonding straps between the metallic

frame of the fillet and the wing leading edge ribs, on both the left and right sides. This proposed AD results from fuel system reviews conducted by the manufacturer. We are proposing this AD to reduce the potential of ignition sources inside fuel tanks in the event of a severe lightning strike, which, in combination with flammable fuel vapors, could result in fuel tank explosions and consequent loss of the airplane.

DATES: We must receive comments on this proposed AD by July 3, 2006.

ADDRESSES: Use one of the following addresses to submit comments on this proposed AD.

- DOT Docket Web site: Go to <http://dms.dot.gov> and follow the instructions for sending your comments electronically.

- Government-wide rulemaking Web site: Go to <http://www.regulations.gov> and follow the instructions for sending your comments electronically.

- Mail: Docket Management Facility, U.S. Department of Transportation, 400 Seventh Street, SW., Nassif Building, room PL-401, Washington, DC 20590.

- Fax: (202) 493-2251.

- Hand Delivery: Room PL-401 on the plaza level of the Nassif Building, 400 Seventh Street, SW., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

Contact Boeing Commercial Airplanes, Long Beach Division, 3855 Lakewood Boulevard, Long Beach, California 90846, Attention: Data and Service Management, Dept. C1-L5A (D800-0024), for the service information identified in this proposed AD.

FOR FURTHER INFORMATION CONTACT:

Samuel Lee, Aerospace Engineer, Propulsion Branch, ANM-140L, FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California 90712-4137; telephone (562) 627-5262; fax (562) 627-5210.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to submit any relevant written data, views, or arguments regarding this proposed AD. Send your comments to an address listed in the **ADDRESSES** section. Include the docket number "FAA-2006-24780; Directorate Identifier 2006-NM-069-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of the proposed AD. We will consider all comments received by the closing date and may amend the proposed AD in light of those comments.

We will post all comments we receive, without change, to <http://dms.dot.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact with FAA personnel concerning this proposed AD. Using the search function of that Web site, anyone can find and read the comments in any of our dockets, including the name of the individual who sent the comment (or signed the comment on behalf of an association, business, labor union, etc.). You may review DOT's complete Privacy Act Statement in the **Federal Register** published on April 11, 2000 (65 FR 19477-78), or you may visit <http://dms.dot.gov>.

Examining the Docket

You may examine the AD docket on the Internet at <http://dms.dot.gov>, or in person at the Docket Management Facility office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The Docket Management Facility office (telephone (800) 647-5227) is located on the plaza level of the Nassif Building at the DOT street address stated in the **ADDRESSES** section. Comments will be available in the AD docket shortly after the Docket Management System receives them.

Discussion

The FAA has examined the underlying safety issues involved in fuel tank explosions on several large transport airplanes, including the adequacy of existing regulations, the service history of airplanes subject to those regulations, and existing maintenance practices for fuel tank systems. As a result of those findings, we issued a regulation titled "Transport Airplane Fuel Tank System Design Review, Flammability Reduction and Maintenance and Inspection Requirements" (67 FR 23086, May 7, 2001). In addition to new airworthiness standards for transport airplanes and new maintenance requirements, this rule included Special Federal Aviation Regulation No. 88 ("SFAR 88," Amendment 21-78, and subsequent Amendments 21-82 and 21-83).

Among other actions, SFAR 88 requires certain type design (i.e., type certificate (TC) and supplemental type certificate (STC)) holders to substantiate that their fuel tank systems can prevent ignition sources in the fuel tanks. This requirement applies to type design holders for large turbine-powered transport airplanes and for subsequent modifications to those airplanes. It requires them to perform design reviews and to develop design changes and

maintenance procedures if their designs do not meet the new fuel tank safety standards. As explained in the preamble to the rule, we intended to adopt airworthiness directives to mandate any changes found necessary to address unsafe conditions identified as a result of these reviews.

In evaluating these design reviews, we have established four criteria intended to define the unsafe conditions associated with fuel tank systems that require corrective actions. The percentage of operating time during which fuel tanks are exposed to flammable conditions is one of these criteria. The other three criteria address the failure types under evaluation: single failures, single failures in combination with a latent condition(s), and in-service failure experience. For all four criteria, the evaluations included consideration of previous actions taken that may mitigate the need for further action.

We have determined that the actions identified in this AD are necessary to reduce the potential of ignition sources inside fuel tanks, which, in combination with flammable fuel vapors, could result in fuel tank explosions and consequent loss of the airplane.

Engineering review of the extended wing-to-fuselage fillet on certain McDonnell Douglas Model DC-10-10, DC-10-10F, DC-10-15, DC-10-30, DC-10-30F (KC-10A and KDC-10), DC-10-40, and DC-10-40F airplanes revealed an increase in the nonmetallic area of the fillet. Engineering reviews of the conventional wing-to-fuselage fillet on certain of the same airplane models revealed that the support ribs of the fuselage-mounted fillet are not grounded, but should be. These conditions, in combination with a severe lightning strike and flammable fuel vapors, could result in fuel tank explosions and consequent loss of the airplane.

Relevant Service Information

We have reviewed McDonnell Douglas DC-10 Service Bulletin 53-109, Revision 4, dated October 7, 1992 (for airplanes with extended wing-to-fuselage fillets); and McDonnell Douglas DC-10 Service Bulletin 53-111, Revision 3, dated August 24, 1992 (for airplanes with conventional wing-to-fuselage fillets). The service bulletins describe procedures for installing or replacing with improved parts, as applicable, the bonding straps between the metallic frame of the fillet and the wing leading edge ribs, on both the left and right sides. For airplanes with extended wing-to-fuselage fillets, the service bulletin indicates that there are

six bonding straps. For airplanes with conventional wing-to-fuselage fillets, the service bulletin indicates that there are ten bonding straps. Accomplishing the actions specified in the service information is intended to adequately address the unsafe condition.

FAA's Determination and Requirements of the Proposed AD

We have evaluated all pertinent information and identified an unsafe condition that is likely to exist or develop on other airplanes of this same type design. For this reason, we are proposing this AD, which would require accomplishing the actions specified in the service information described previously, except as discussed under "Difference Between the Proposed AD and the Service Bulletins."

Difference Between the Proposed AD and the Service Bulletins

McDonnell Douglas DC-10 Service Bulletin 53-109 recommends doing the installation or replacement at the earliest practical maintenance period, and McDonnell Douglas DC-10 Service Bulletin 53-111 recommends doing the installation or maintenance at the first convenient check, but no later than 7,500 flight-hours after receiving the service bulletin. We have determined that these intervals would not address the identified unsafe condition soon enough to ensure an adequate level of safety for the affected fleet. In developing an appropriate compliance time for this AD, we considered the manufacturer's recommendation, the degree of urgency associated with the subject unsafe condition, and the average utilization of the affected fleet. In light of all of these factors, we find that a compliance time of the earlier of 7,500 flight hours or 60 months after the effective date of this AD represents an appropriate interval of time for affected airplanes to continue to operate without compromising safety. This difference has been coordinated with Boeing, and Boeing concurs.

Costs of Compliance

There are about 457 airplanes of the affected design in the worldwide fleet. This proposed AD would affect about 280 airplanes of U.S. registry. The proposed actions would take between 9 and 17 work hours per airplane, at an average labor rate of \$80 per work hour. Required parts would cost between \$3,720 and \$4,169 per airplane. Based on these figures, the estimated cost of the proposed AD is between \$4,440 and \$5,529 per airplane, or between \$1,243,200 and \$1,548,120 for the U.S.-registered fleet.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We have determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that the proposed regulation:

1. Is not a "significant regulatory action" under Executive Order 12866;

2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and

3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this proposed AD and placed it in the AD docket. See the **ADDRESSES** section for a location to examine the regulatory evaluation.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. The Federal Aviation Administration (FAA) amends § 39.13 by adding the following new airworthiness directive (AD):

McDonnell Douglas: Docket No. FAA-2006-24780; Directorate Identifier 2006-NM-069-AD.

Comments Due Date

(a) The FAA must receive comments on this AD action by July 3, 2006.

Affected ADs

(b) None.

Applicability

(c) This AD applies to McDonnell Douglas Model DC-10-10, DC-10-10F, DC-10-15, DC-10-30, DC-10-30F (KC-10A and KDC-10), DC-10-40, DC-10-40F, airplanes, certificated in any category; as identified in the applicable service bulletin listed in Table 1 of this AD.

TABLE 1.—SERVICE BULLETINS

McDonnell Douglas DC-10 service bulletin	Revision level	Date	For airplanes with—
53-109	4	October 7, 1992	Extended wing-to-fuselage fillets. Conventional wing-to-fuselage fillets.
53-111	3	August 24, 1992	

Unsafe Condition

(d) This AD results from fuel system reviews conducted by the manufacturer. We are issuing this AD to reduce the potential of ignition sources inside fuel tanks in the event of a severe lightning strike, which, in combination with flammable fuel vapors, could result in fuel tank explosions and consequent loss of the airplane.

Compliance

(e) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

Installation or Replacement

(f) Within 7,500 flight hours or 60 months after the effective date of this AD, whichever occurs earlier: Install or replace with improved parts, as applicable, the bonding straps between the metallic frame of the fillet and the wing leading edge ribs, on both the left and right sides, in accordance with the Accomplishment Instructions of the applicable service bulletin identified in Table 1 of this AD.

Alternative Methods of Compliance (AMOCs)

(g)(1) The Manager, Los Angeles Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested in accordance with the procedures found in 14 CFR 39.19.

(2) Before using any AMOC approved in accordance with § 39.19 on any airplane to which the AMOC applies, notify the appropriate principal inspector in the FAA Flight Standards Certificate Holding District Office.

Issued in Renton, Washington, on May 8, 2006.

Ali Bahrami,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. E6-7476 Filed 5-16-06; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA-2006-24787; Directorate Identifier 2006-NM-043-AD]

RIN 2120-AA64

Airworthiness Directives; McDonnell Douglas Model DC-10-10 and DC-10-10F Airplanes; Model DC-10-15 Airplanes; Model DC-10-30 and DC-10-30F (KC-10A and KDC-10) Airplanes; Model DC-10-40 and DC-10-40F Airplanes; Model MD-10-10F and MD-10-30F Airplanes; and Model MD-11 and MD-11F Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: The FAA proposes to adopt a new airworthiness directive (AD) for certain McDonnell Douglas transport category airplanes. This proposed AD would require fabrication and installation of a wire harness guard in the right wheel well of the main landing gear (MLG), and related investigative and corrective actions as necessary. For certain airplanes, the proposed AD also would require replacement of the electrical connectors of the auxiliary hydraulic pumps with improved electrical connectors and related investigative and corrective actions. This proposed AD results from fuel system reviews conducted by the manufacturer. We are proposing this AD to prevent damage to the wire support bracket and wiring of the auxiliary hydraulic pump and, for certain airplanes, water intrusion through the electrical connectors of the auxiliary hydraulic pump. These conditions could lead to a potential ignition source in the right wheel well of the MLG around the fuel tank, which, in combination with flammable fuel vapors, could result in fuel tank explosions and consequent loss of the airplane.

DATES: We must receive comments on this proposed AD by July 3, 2006.

ADDRESSES: Use one of the following addresses to submit comments on this proposed AD.

- DOT Docket Web site: Go to <http://dms.dot.gov> and follow the instructions for sending your comments electronically.

- Government-wide rulemaking Web site: Go to <http://www.regulations.gov> and follow the instructions for sending your comments electronically.

- Mail: Docket Management Facility, U.S. Department of Transportation, 400 Seventh Street, SW., Nassif Building, room PL-401, Washington, DC 20590.
- Fax: (202) 493-2251.
- Hand Delivery: Room PL-401 on the plaza level of the Nassif Building, 400 Seventh Street, SW., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

Contact Boeing Commercial Airplanes, Long Beach Division, 3855 Lakewood Boulevard, Long Beach, California 90846, Attention: Data and Service Management, Dept. C1-L5A (D800-0024), for the service information identified in this proposed AD.

FOR FURTHER INFORMATION CONTACT: Ken Sujishi, Aerospace Engineer, Cabin Safety/Mechanical and Environmental Systems Branch, ANM-150L, FAA, Los

Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California 90712-4137; telephone (562) 627-5353; fax (562) 627-5210.

SUPPLEMENTARY INFORMATION:**Comments Invited**

We invite you to submit any relevant written data, views, or arguments regarding this proposed AD. Send your comments to an address listed in the **ADDRESSES** section. Include the docket number "FAA-2006-24787; Directorate Identifier 2006-NM-043-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of the proposed AD. We will consider all comments received by the closing date and may amend the proposed AD in light of those comments.

We will post all comments we receive, without change, to <http://dms.dot.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact with FAA personnel concerning this proposed AD. Using the search function of that Web site, anyone can find and read the comments in any of our dockets, including the name of the individual who sent the comment (or signed the comment on behalf of an association, business, labor union, etc.). You may review DOT's complete Privacy Act Statement in the **Federal Register** published on April 11, 2000 (65 FR 19477-78), or you may visit <http://dms.dot.gov>.

Examining the Docket

You may examine the AD docket on the Internet at <http://dms.dot.gov>, or in person at the Docket Management Facility office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The Docket Management Facility office (telephone (800) 647-5227) is located on the plaza level of the Nassif Building at the DOT street address stated in the **ADDRESSES** section. Comments will be available in the AD docket shortly after the Docket Management System receives them.

Discussion

The FAA has examined the underlying safety issues involved in fuel tank explosions on several large transport airplanes, including the adequacy of existing regulations, the service history of airplanes subject to those regulations, and existing maintenance practices for fuel tank systems. As a result of those findings, we issued a regulation titled "Transport Airplane Fuel Tank System Design