

Rules does not affect the collections of information outlined in the Interim Rule nor does it affect the estimated burden set forth in the Interim Rule.

List of Subjects in 12 CFR Part 370

Banks, Banking, Bank deposit insurance, Holding companies, National banks, Reporting and recordkeeping requirements, Savings associations.

■ For the reasons stated above, The Board of Directors of the Federal Deposit Insurance Corporation amends 12 CFR part 370 as follows:

PART 370—TEMPORARY LIQUIDITY GUARANTEE PROGRAM

■ 1. The authority citation for part 370 continues to read as follows:

Authority: 12 U.S.C. 1813(l), 1813(m), 1817(i), 1818, 1819(a) (Tenth); 1820(f), 1821(a); 1821(c); 1821(d); 1823(c)(4).

§ 370.2 [Amended]

■ 2. Amend § 370.2 as follows:

■ A. In paragraph (f), remove “November 12” and replace it with “December 5”.

■ B. In paragraph (g), remove “November 12” and replace it with “December 5” and remove “November 13” wherever it appears and replace it with “December 6”.

§ 370.3 [Amended]

■ 3. Amend § 370.3 as follows:

■ In paragraphs (b) and (f), remove “November 12” wherever it appears and replace it with “December 5”.

§ 370.5 [Amended]

■ 4. Amend § 370.5 as follows:

■ A. In paragraphs (a), (c), (f), and (j), remove “November 12” wherever it appears and replace it with “December 5”.

■ B. In paragraph (h), remove “December 1” and replace it with “December 19”.

■ 5. Amend § 370.6 by revising paragraphs (a), (b)(1), (b)(2), and (c) to read as follows:

§ 370.6 Assessments under the Debt Guarantee Program.

(a) *Waiver of assessment for certain initial periods.* No eligible entity shall pay any assessment associated with the debt guarantee program for the period from October 14, 2008 through November 12, 2008. An eligible entity that opts out of the program on or before December 5, 2008 will not pay any assessment under the program.

(b) * * *

(1) Any eligible entity that does not opt out of the Debt Guarantee Program on or before December 5, 2008, as

provided in § 370.5, and that issues any guaranteed debt during the period from October 14, 2008 through December 5, 2008 which is still outstanding on December 5, 2008, shall notify the FDIC of that issuance via the FDIC’s e-business Web site *FDICconnect* on or before December 19, 2008, and the eligible entity’s Chief Financial Officer or equivalent shall certify that the issuances outstanding at each point of time did not exceed the guaranteed amount limit as set forth in § 370.3.

(2) Any eligible entity that does not opt out of the program and that issues guaranteed debt after December 5, 2008, shall notify the FDIC of that issuance via the FDIC’s e-business Web site *FDICconnect* within the time period specified by the FDIC. The eligible entity’s Chief Financial Officer or equivalent shall certify that the issuance of guaranteed debt does not exceed the guarantee limit as set forth in § 370.3.

* * * * *

(c) *Initiation of assessments.*

Assessments, calculated in accordance with paragraph (d) of this section, will accrue, with respect to each eligible entity that does not opt out of the debt guarantee program on or before December 5, 2008.

(1) Beginning on November 13, 2008, on all senior unsecured debt, other than overnight debt instruments, issued by it on or after October 14, 2008 that is still outstanding on November 13, 2008;

(2) Beginning on November 13, 2008, on all senior unsecured debt, other than overnight debt instruments, issued by it on or after November 13, 2008 and before December 6, 2008; and

(3) Beginning on December 6, 2008, on all senior unsecured debt issued by it on or after December 6, 2008.

■ 6. Amend § 370.7 by revising paragraphs (a) and (b) to read as follows:

§ 370.7 Assessments under the Transaction Account Guarantee Program.

(a) *Waiver of assessment for certain initial periods.* No eligible entity shall pay any assessment associated with the transaction account guarantee program for the period from October 14, 2008, through November 12, 2008. An eligible entity that opts out of the program on or before December 5, 2008 will not pay any assessment under the program

(b) *Initiation of assessments.* Beginning on November 13, 2008 each eligible entity that does not opt out of the transaction account guarantee program on or before December 5, 2008 will be required to pay the FDIC assessments on all deposit amounts in noninterest-bearing transaction accounts

calculated in accordance with paragraph (c) of this section.

* * * * *

Federal Deposit Insurance Corporation.

Robert E. Feldman,

Executive Secretary.

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 23

[Docket No. CE285, Special Conditions No. 23–225A–SC]

Final Special Conditions: AmSafe Aviation; Inflatable Restraints Installation; Approved Model List of Normal and Utility Category Airplanes, and Agricultural Airplanes Certified in the Normal/Utility/Restricted Category

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final special conditions; request for comments.

SUMMARY: These special conditions are issued for AmSafe Aviation to amend the list of approved models. These airplanes, as modified by AmSafe Aviation, will have novel and unusual design features associated with the lap belt or shoulder harness portion of the safety belt, which contains an integrated airbag device. The applicable airworthiness regulations do not contain adequate and appropriate safety standards for this design feature. These special conditions contain the additional safety standards that the Administrator considers necessary to establish a level of safety equivalent to that established by the airworthiness standards.

DATES: The effective date of these amended special conditions is October 31, 2008. Comments must be received on or before December 8, 2008.

ADDRESSES: Mail two copies of your comments on these amended special conditions to: Federal Aviation Administration (FAA), Regional Counsel, ACE–7, Attention: Rules Docket, Docket No. CE285, 901 Locust, Room 506, Kansas City, Missouri 64106, or you may deliver two copies to the Regional Counsel at the above address. Mark your comments: Docket No. CE285. You may inspect comments in the Rules Docket weekdays, except Federal holidays, between 7:30 a.m. and 4 p.m.

FOR FURTHER INFORMATION CONTACT: Mr. Bob Stegeman, Federal Aviation Administration, Small Airplane Directorate, Aircraft Certification Service, ACE-111, 901 Locust, Kansas City, Missouri, 816-329-4140, fax 816-329-4090, e-mail Robert.Stegeman@faa.gov.

SUPPLEMENTARY INFORMATION: The FAA has determined that notice and opportunity for prior public comment hereon are impracticable because these procedures would significantly delay issuance of the approval design and thus delivery of the affected aircraft. In addition, the substance of these special conditions has been subject to the public comment process in several prior instances with no substantive comments received. The FAA, therefore, finds that good cause exists for making these amended special conditions effective upon issuance.

Comments Invited

We invite interested persons to participate in the making of these special conditions by submitting such written data, views, or arguments as they may desire. Identify the regulatory docket or notice number and submit the comments in duplicate to the address specified above. All comments received on or before the closing date for comments will be considered by the Administrator. The special conditions may be changed in light of the comments received. All comments received will be available in the Rules Docket for examination by interested persons, both before and after the closing date for comments. A report summarizing each substantive public contact with FAA personnel concerning this rulemaking will be filed in the docket. Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this notice must include a self-addressed, stamped postcard on which the following statement is made: "Comments to CE285." The postcard will be date stamped and returned to the commenter.

Background

On June 18, 2008, AmSafe Aviation, 1043 North 47th Avenue, Phoenix, AZ 85043, applied to add several more models to an existing supplemental type certificate. The AML STC allows for the installation of inflatable restraints in airplane models included herein that were certificated prior to the dynamic seat rule specified in 14 CFR part 23, § 23.562.

AmSafe Aviation has previously applied for and obtained an Approved

Model List (AML) Supplemental Type Certificate (STC) for the installation of Inflatable Two-, Three-, Four- or Five-Point Restraint Safety Belts with an Integrated Airbag Device in airplanes certificated in the Part 23 Normal/Utility and Restricted-Agricultural categories.

This special condition includes additional normal and utility category aircraft.

The inflatable restraint system is either a two-, three-, four-, or five-point safety belt restraint system consisting of a shoulder harness and a lap belt with an inflatable airbag attached to either the lap belt or the shoulder harness. The inflatable portion of the restraint system will rely on sensors to electronically activate the inflator for deployment. The inflatable restraint system will be made available on the pilot, co-pilot, and passenger seats of these airplanes.

If an emergency landing occurs, the airbag will inflate and provide a protective cushion between the occupant's head and structure within the airplane. This will reduce the potential for head and torso injury. The inflatable restraint behaves in a manner that is similar to an automotive airbag. However, in this case, the airbag is integrated into the lap or shoulder belt. While airbags and inflatable restraints are standard in the automotive industry, the use of an inflatable restraint system is novel for aircraft operations.

The FAA has determined that this project will be accomplished on the basis of not lowering the current level of safety of the airplanes' original certification basis. The FAA has two primary safety concerns with the installation of airbags or inflatable restraints:

- That they perform properly under foreseeable operating conditions; and
- That they do not perform in a manner or at such times as to impede the pilot's ability to maintain control of the airplane or constitute a hazard to the airplane or occupants.

The latter point has the potential to be the more rigorous of the requirements. An unexpected deployment while conducting the takeoff or landing phases of flight may result in an unsafe condition. The unexpected deployment may either startle the pilot, or it may generate a force sufficient to cause a sudden movement of the control yoke. Either action could result in a loss of control of the airplane, the consequences of which are magnified due to the low operating altitudes during these phases of flight. This consideration is of special concern for aircraft designated for agricultural use because these aircraft spend a majority

of their flight time at low altitudes. The FAA has considered this when establishing these amended special conditions.

The inflatable restraint system relies on sensors to electronically activate the inflator for deployment. These sensors could be susceptible to inadvertent activation, causing deployment in a potentially unsafe manner. The consequences of an inadvertent deployment must be considered in establishing the reliability of the system. AmSafe Aviation must show that the effects of an inadvertent deployment in flight are not a hazard to the airplane or that an inadvertent deployment is extremely improbable.

In addition, general aviation and agricultural aircraft are susceptible to a large amount of cumulative wear and tear on a restraint system. It is likely that the potential for inadvertent deployment increases as a result of this cumulative damage. Therefore, the impact of wear and tear on inadvertent deployment must be considered. Due to the effects of this cumulative damage, a life limit must be established for the appropriate system components in the restraint system design.

There are additional factors to be considered to minimize the chances of inadvertent deployment. General aviation airplanes are exposed to a unique operating environment, since the same airplane may be used by both experienced and student pilots. The effect of this environment on inadvertent deployment must be understood. Therefore, qualification testing of the firing hardware/software must consider the following:

- The airplane vibration levels appropriate for general aviation and agricultural airplanes; and
- The inertial loads that result from typical flight/ground maneuvers, gusts, hard landings and flight maneuvering unique to both general aviation and agricultural aircraft operations.

Any tendency for the firing mechanism to activate as a result of these loads or acceleration levels is unacceptable.

Other influences on inadvertent deployment include high intensity electromagnetic fields (HIRF) and lightning. Since the sensors that trigger deployment are electronic, they must be protected from the effects of these threats. To comply with HIRF and lightning requirements, the AmSafe Aviation inflatable restraint system is considered a critical system, since its inadvertent deployment could have a hazardous effect on the airplane.

Given the level of safety of the retrofitted airplane occupant restraints,

the inflatable restraint system must show that it will offer an equivalent level of protection in the event of an emergency landing. If a deployment occurs, the restraint must still be at least as strong as a Technical Standard Order approved belt and shoulder harnesses. There is no requirement for the inflatable portion of the restraint to offer protection during multiple impacts.

The inflatable restraint system must deploy and provide protection for each occupant during crash conditions as specified in the original certification basis. Therefore, the test emergency landing loads identified in the original certification basis of the airplane must be used to satisfy this requirement. It must be shown that the inflatable restraint will deploy and provide protection under crash conditions as specified in the original certification basis. Compliance will be demonstrated using the test condition specified in the original certification basis. It must be shown that the crash sensor will trigger when exposed to a rapidly applied deceleration, like an actual crash event. Therefore, the test crash pulses identified in § 23.562 must be used to satisfy this requirement, although the peak “G” may be reduced to a level meeting the original certification requirements of the aircraft. Testing to these pulses will demonstrate that the crash sensor will trigger when exposed to a rapidly applied deceleration, like an actual crash event.

It is possible a wide range of occupants will use the inflatable restraint. Thus, the protection offered by this restraint should be effective for occupants that range from the fifth percentile female to the ninety-fifth percentile male.

In support of this operational capability, there must be a means to verify the integrity of this system before each flight. As an option, AmSafe Aviation can establish inspection intervals where they have demonstrated

the system to be reliable between these intervals.

An inflatable restraint may be “armed” even though no occupant is using the seat. While there will be means to verify the integrity of the system before flight, it is also prudent to require that unoccupied seats with active restraints not constitute a hazard to any occupant. This will protect any individual performing maintenance inside the cockpit while the aircraft is on the ground. The restraint must also provide suitable visual warnings that would alert rescue personnel to the presence of an inflatable restraint system.

In addition, the design must prevent the inflatable seatbelt from being incorrectly buckled and/or installed such that the airbag would not properly deploy. As an alternative, AmSafe Aviation may show that such deployment is not hazardous to the occupant and will still provide the required protection.

The cabins of the various model airplanes identified in these special conditions are confined areas, and the FAA is concerned that noxious gases may accumulate if an airbag deploys. When deployment does occur, either by design or inadvertently, there must not be a release of hazardous quantities of gas or particulate matter into the cockpit.

An inflatable restraint should not increase the risk already associated with fire. Therefore, the inflatable restraint should be protected from the effects of fire, so that an additional hazard is not created by, for example, a rupture of the inflator.

Finally, the airbag is likely to have a large volume displacement, and it may impede the egress of an occupant. Since the bag deflates to absorb energy, it is likely that the inflatable restraint would be deflated at the time an occupant would attempt egress. However, it is appropriate to specify a time interval after which the inflatable restraint may not impede rapid egress. Ten seconds

has been chosen as reasonable time. This time limit will offer a level of protection throughout the impact event.

Special conditions for the installation of AAIR systems on other certificated airplanes have been issued and no substantive public comments were received. Since the same special conditions were issued multiple times for different model airplanes with no substantive public comments, the FAA began issuing direct final special conditions with an invitation for public comment. This was done to eliminate the waiting period for public comments and to allow AmSafe Aviation to proceed with the project.

These previous special conditions were typically issued for a single model airplane or for variants of a model from a single airplane manufacturer, and required dynamic testing of each AAIR system installation for showing compliance. Since AmSafe Aviation has previously demonstrated by dynamic testing, and has the supporting data, that the Electronics Module Assembly (EMA) and inflator assembly will function as intended in a simulated dynamic emergency landing, it is not necessary to repeat the test for each airplane model shown in these amended special conditions.

Type Certification Basis

Under the provisions of 14 CFR part 21, § 21.101, AmSafe Aviation must show that affected airplane models, as changed, continue to meet the applicable provisions of the regulations incorporated by reference in the Type Certificate Numbers listed below or the applicable regulations in effect on the date of application for the change. The regulations incorporated by reference in the type certificate data sheet are commonly referred to as the original “type certification basis” and can be found in the Type Certificate Data Sheets for the numbers listed below. The following models are covered by this amended special condition:

LIST OF ALL AIRPLANE MODELS AND APPLICABLE TCDS

Make	Model	TC holder	TCDS	Certification basis
1 Aerostar	PA-60-600 (Aerostar 600) PA-60-601 (Aerostar 601) PA-60-601P (Aerostar 601P) PA-60-602P (Aerostar 602P) PA-60-700P (Aerostar 700P)	Aerostar Aircraft Corporation.	A17WE, Revision 22	14 CFR PART 23.
1 All American	10A	All American Aircraft, Inc	A-792	CAR 3.
1 American Champion (Champion).	402	American Champion Aircraft Corp.	A3CE, Revision 5	CAR 3.

LIST OF ALL AIRPLANE MODELS AND APPLICABLE TCDS—Continued

Make	Model	TC holder	TCDS	Certification basis
1 American Champion (Bellanca) (Champion) (Aeronca).	7AC, 7ACA, 7EC, 7GCB, S7AC, S7EC, 7GCBA (L-16A), 7BCM, 7ECA, 7GCBC (L-16B), 7CCM, 7FC, 7HC, S7CCM, 7GC, 7JC, 7DC, 7GCA, 7KC, S7DC, 7GCAA, 7KCAB.	American Champion Aircraft Corp.	A-759, Revision 67	CAR 4a.
1 American Champion (Bellanca) (Trytek) (Aeronca).	11AC, S11AC, 11BC, S11BC	American Champion Aircraft Corp.	A-761, Revision 17	CAR 4a.
1 American Champion (Bellanca) (Trytek) (Aeronca).	11CC, S11CC	American Champion Aircraft Corporation.	A-796, Revision 14	CAR 3.
1 VARGA (Morrisey)	2150, 2150A, 2180	Augustair, Inc.	4A19, Revision 9	CAR 3.
1 Bellanca	14-13, 14-13-2, 14-13-3, 14-13-3W	Bellanca Aircraft Corporation.	A-773, Revision 10	CAR 4a.
1 Bellanca	14-9, 14-9L	Bellanca Aircraft Corporation.	TC716	CAR 4a.
1 Cessna	120, 140	Cessna Aircraft Company.	A-768, Revision 34	CAR 4a.
1 Cessna	140A	Cessna Aircraft Company.	5A2, Revision 21	CAR 3.
1 Cessna	150, 150J, 150A, 150K, 150B, A150K, 150C, 150L, 150D, A150L, 150E, 150M, 150F, A150M, 150G, 152, 150H, A152.	Cessna Aircraft Company.	3A19, Revision 44	CAR 3.
1 Cessna	170, 170A, 170B	Cessna Aircraft Company.	A-799, Revision 54	CAR 3.
1 Cessna	172, 172I, 172A, 172K, 172B, 172L, 172C, 172M, 172D, 172N, 172E, 172P, 172F (USAF T-41A), 172Q, 172G, 172H, (USAF T-41A).	Cessna Aircraft Company.	3A12, Revision 73	CAR 3.
1 Cessna	175, 175A, 175B, 175C, P172D, R172E (USAF T-41B) (USAF T-41C and D), R172F (USAF T-41D), R172G (USAF T-41C or D), R172H (USAF T-41D), R172J, R172K, 172RG.	Cessna Aircraft Company.	3A17, Revision 45	CAR 3.
1 Cessna	177, 177A, 177B	Cessna Aircraft Company.	A13CE, Revision 24	14 CFR PART 23.
Cessna	177RG	Cessna Aircraft Company.	A20CE, Revision 21	14 CFR PART 23.
Cessna	180, 180E, 180A, 180F, 180B, 180G, 180C, 180H, 180D, 180J, 180E, 180K.	Cessna Aircraft Company.	5A6, Revision 66	CAR 3.
1 Cessna	182, 182K, 182A, 182L, 182B, 182M, 182C, 182N, 182D, 182P, 182E, 182Q, 182F, 182R, 182G, R182, 182H, T182, 182J, TR182.	Cessna Aircraft Company.	3A13, Revision 64	CAR 3.
1 Cessna	185, A185E, 185A, A185F, 185B, 185C, 185D, 185E.	Cessna Aircraft Company.	3A24, Revision 37	CAR 3.
Cessna AgWagon	188, 188A, 188B, A188, A188A, A188B, T188C.	Cessna Aircraft Company.	A9CE, Revision 27	14 CFR PART 23.
1 Cessna	190 (LC-126A,B,C), 195, 195A, 195B	Cessna Aircraft Company.	A-790, Revision 36	CAR 3.
1 Cessna	206, U206B, TP206D, P206, U206C, TP206E, P206A, U206D, TU206A, P206B, U206E, TU206B, P206C, U206F, TU206C, P206D, U206G, TU206D, P206E, TP206A, TU206E, U206, TP206B, TU206F, U206A, TP206C, TU206G.	Cessna Aircraft Company.	A4CE, Revision 43	CAR 3.
1 Cessna	208, 208A, 208B	Cessna Aircraft Company.	A37CE, Revision 12	14 CFR PART 23.
1 Cessna	210, 210K, 210A, T210K, 210B, 210L, 210C, T210L, 210D, 210M, 210E, T210M, 210F, 210N, T210F, P210N, 210G, T210N, T210G, 210R, 210H, P210R, T210H, T210R, 210J, 210-5 (205), T210J, 210-5A (205A).	Cessna Aircraft Company.	3A21, Revision 46	CAR 3.
1 Cessna	310, 310J, 310A(USAF U-3A), 310J-1, 310B, E310J, 310C, 310K, 310D, 310L, 310E(USAF U-3B), 310N, 310F, 310P, 310G, T310P, 310H, 310Q, E310H, T310Q, 310I, 310R, T310R.	Cessna Aircraft Company.	3A10, Revision 62	CAR 3.

LIST OF ALL AIRPLANE MODELS AND APPLICABLE TCDS—Continued

Make	Model	TC holder	TCDS	Certification basis
1 Cessna	320, 320F, 320-1, 335, 320A, 340, 320B, 340A, 320C, 320D, 320E.	Cessna Aircraft Company.	3A25, Revision 25	CAR 3.
1 Cessna	321 (Navy OE-2)	Cessna Aircraft Company.	3A11, Revision 6	CAR 3.
1 Cessna	336	Cessna Aircraft Company.	A2CE, Revision 7	CAR 3.
1 Cessna	337A (USAF 02B), T337E, 337B, 337F, M337B (USAF 02A), T337F, T337B, 337G, 337C, T337G, T337C, 337H, 337D, P337H, T337D, T337H, T337H-SP.	Cessna Aircraft Company.	A6CE, Revision 40	CAR 3/14 CFR PART 23.
1 Cessna	401, 401A, 401B, 402, 402A, 402B, 402C, 411, 411A, 414, 414A, 421, 421A, 421B, 421C, 425.	Cessna Aircraft Company.	A7CE, Revision 46	CAR 3.
1 Cessna	404, 406	Cessna Aircraft Company.	A25CE, Revision 11	14 CFR PART 23.
1 Cessna	441	Cessna Aircraft Company.	A28CE, Revision 12	14 CFR PART 23.
1 Commander Aircraft	Model 112, Model 114, Model 112TC, Model 112B, Model 112TCA, Model 114A, Model 114B, Model 114TC.	Commander Aircraft Company.	A12SO, Revision 21	14 CFR PART 23.
Diamond	DA20-A1, DA20-C1	Diamond Aircraft Industries, Inc..	TA4CH, Revision 14	14 CFR PART 23.
1 Great Lakes	2T-1A, 2T-1A-1, 2T-1A-2	Great Lakes Aircraft Company, LLC.	A18EA, Revision 10	Aeronautical Bulletin No. 7-A.
1 Helio (Taylorcraft)	15A, 20	Helio Aircraft Corporation.	3A3, Revision 7	CAR 4a.
1 Learjet	23	Learjet Inc	A5CE, Revision 10	CAR 3.
1 Lockheed	402-2	Lockheed Aircraft International.	2A11, Revision 4	CAR 3.
1 Land-Air (TEMCO) (Luscombe).	11A, 11E	Luscombe Aircraft Corporation.	A-804, Revision 14	CAR 3.
1 Maule	Bee Dee M-4, M-5-180C, MXT-7-160, M-4-180V, M-4 M-5-200, MX-7-180A, M-4C, M-5-210C, MXT-7-180A, M-4S, M-5-210TC, MX-7-180B, M-4T, M-5-220C, M-7-235B, M-4-180C, M-5-235C, M-7-235A, M-4-180S, M-6-180, M-7-235C, M-4-180T, M-6-235, MX-7-180C, M-4-210, M-7-235, M-7-260, M-4-210C, MX-7-235, MT-7-260, M-4-210S, MX-7-180, M-7-260C, M-4-210T, MX-7-420, M-7-420AC, M-4-220, MXT-7-180, MX-7-160C, M-4-220C, MT-7-235, MX-7-180AC, M-4-220S, M-8-235, M-7-420A, M-4-220T, MX-7-160, MT-7-420.	Maule Aerospace Technology, Inc.	3A23, Revision 30	CAR 3.
1 Mooney	M20, M20A, M20B, M20C, M20D, M20E, M20F, M20G, M20J, M20K (Up to S/N 25-2000), M20L.	Mooney Airplane Company, Inc.	2A3, Revision 47	CAR 3.
1 Interceptor (Aero Commander) (Meyers).	200, 200A, 200B, 200C, 200D, 400	Prop-Jets, Inc	3A18, Revision 16	CAR 3.
1 Beech	35-33, J35, 35-A33, K35, 35-B33, M35, 35-C33, N35, 35-C33A, P35, E33, S35, E33A, V35, E33C, V35A, F33, V35B, F33A, 36, F33C, A36, G33, A36TC, H35, B36TC, G36.	Raytheon Aircraft Company.	3A15, Revision 90	CAR 3.
1 Beech	45 (YT-34), A45 (T-34A, B-45), D45 (T-34B).	Raytheon Aircraft Company.	5A3, Revision 25	CAR 03.
1 Beech	19A, B23, B19, C23, M19A, A24, 23, A24R, A23, B24R, A23A, C24R, A23-19, A23-24.	Raytheon Aircraft Company.	A1CE, Revision 34	CAR 3.
1 Beech	3N, 3NM, 3TM, JRB-6, D18C, D18S, E18S, E18S-9700, G18S, H18, C-45G, TC-45G, C-45H, TC-45H, TC-45J or UC-45J (SNB-5), RC-45J (SNB-5P).	Raytheon Aircraft Company.	A-765	Revision 74 CAR 03.
1 Beech	35, A35, E35, B35, F35, C35, G35, D35, 35R.	Raytheon Aircraft Company.	A-777, Revision 57	CAR 03.

LIST OF ALL AIRPLANE MODELS AND APPLICABLE TCDS—Continued

Make	Model	TC holder	TCDS	Certification basis
1 Raytheon	200, A100-1 (U-21J), 200C, A200 (C-12A), 200CT, A200 (C-12C), 200T, A200C (UC-12B), B200, A200CT (C-12D), B200C, A200CT (FWC-12D), B200CT, A200CT (C-12F), B200T, A200CT (RC-12D), 300, A200CT (RC-12G), 300LW, A200CT (RC-12H), B300, A200CT (RC-12K), B300C, A200CT (RC-12P), 1900, A200CT (RC-12Q), 1900C, B200C (C-12F), 1900D, B200C (UC-12M), B200C (C-12R), B200C (UC-12F), 1900C (C-12J).	Raytheon Aircraft Company.	A24CE, Revision 91	14 CFR PART 23.
1 Beech	B95A, D55, D95A, D55A, E95, E55, 95-55, E55A, 95-A55, 56TC, 95-B55, A56TC, 95-B55A, 58, 95-B55B (T-42A), 58A, 95-C55, 95, 95-C55A, B95, G58.	Raytheon Aircraft Company.	3A16, Revision 81	CAR 3.
1 Beech	60, A60, B60	Raytheon Aircraft Company.	A12CE, Revision 23	14 CFR PART 23.
1 Beech	58P, 58PA, 58TC, 58TCA	Raytheon Aircraft Company.	A23CE, Revision 14	14 CFR PART 23.
1 Cessna	CESSNA F172D CESSNA F172E CESSNA F172F CESSNA F172G CESSNA F172H CESSNA F172K CESSNA F172L CESSNA F172M CESSNA F172N CESSNA F172P	Reims Aviation S.A.	A4EU, Revision 11	CAR 10/CAR 3.
1 Socata	TB 9, TB 10, TB 20, TB 21, TB 200	SOCATA-GROUPE AEROSPATIALE.	A51EU, Revision 14	14 CFR PART 23.
1 Pitts	S-1S, S-1T, S-2, S-2A, S-2S, S-2B, S-2C.	Sky International Inc. (Aviat Aircraft, Inc.).	A8SO, Revision 21	14 CFR PART 23.
1 Taylorcraft	19, F19, F21, F21A, F21B, F22, F22A, F22B, F22C.	Taylorcraft Aviation, LLC	1A9, Revision 19	CAR 3.
1 Taylorcraft	BC, BCS12-D, BCS, BC12-D1, BC-65, BCS12-D1, BCS-65, BC12D-85, BC12-65 (Army L-2H), BCS12D-85, BCS12-65, BC12D-4-85, BC12-D, BCS12D-4-85.	Taylorcraft Aviation, LLC	A-696, Revision 22	CAR 04.
1 Taylorcraft	(Army L-2G) BF, BFS, BF-60, BFS-60, BF-65, BFS-65, (Army L-2K) BF 12-65, BFS-65.	Taylorcraft, Inc.	A-699, Revision 5	CAR 4a.
1 Luscombe	8, 8D, 8A, 8E, 8B, 8F, 8C, T-8F	The Don Luscombe Aviation History Foundation, Inc.	A-694, Revision 23	CAR 4a.
Sierra Hotel Aero, Inc. (Navion).	Navion (L-17A), Navion A (L-17B) (L-17C), Navion B, Navion D, Navion E, Navion F, Navion G, Navion H.	Sierra Hotel Aero, Inc.	A-782, Revision 51	CAR 3.
Piper	J-3	Piper Aircraft Inc.	ATC 660, Revision 0	Not listed.
Piper	J3C-40, J3C-50, J3C-50S, J3C-65, J3C-65S, PA-11, PA-11S.	Piper Aircraft Inc.	A-691, Revision 33	CAR 4a.
FS 2003 Corporation (Piper).	PA-12, PA-12S	FS 2003 Corporation	A-780, Revision 13	CAR 3.
FS 2002 Corporation (Piper).	PA-14	FS 2002 Corporation	A-797, Revision 11	CAR 3.
Piper	PA-15	Piper Aircraft Inc.	A-800, Revision 11	CAR 3.
Piper	PA-16, PA-16S	Piper Aircraft Inc.	1A1, Revision 13	CAR 3.
Piper	PA-17	Piper Aircraft Inc.	A-805, Revision 12	CAR 3.
2 Piper	PA-18, PA-18S, PA-18A, PA-18S "125", PA-18AS "125", PA-18A "135", PA-18S "135", PA-18AS "135", PA-18 "150", PA-18A "150", PA-18S "150", PA-18AS "150", PA-19S.	The New Piper Aircraft, Inc.	1A2, Revision 37	CAR 3.
Piper	PA-20, PA-20-115, PA-20-135, PA-20S, PA-20S-115, PA-20S-135.	Piper Aircraft Inc.	1A4, Revision 24	CAR 3.

LIST OF ALL AIRPLANE MODELS AND APPLICABLE TCDS—Continued

Make	Model	TC holder	TCDS	Certification basis
Piper	PA-22, PA-22-108, PA-22-135, PA-22-150, PA-22-160, PA-22S-135, PA-22S-150, PA-22S-160.	Piper Aircraft Inc	1A6, Revision 34	CAR 3.
Piper	PA-23, PA-23-160, PA-23-235, PA-23-250.	Piper Aircraft Inc	1A10, Revision 51	CAR 3.
Piper	PA-24, PA-24-250, PA-24-260, PA-24-400.	Piper Aircraft Inc	1A15, Revision 34	CAR 3.
1 Piper	PA-28-140, PA-28-151, PA-28-150, PA-28-161, PA-28-160, PA-28-181, PA-28-180, PA-28R-201, PA-28-235, PA-28R-201T, PA-28S-160, PA-28-236, PA-28S-180, PA-28RT-201, PA-28R-180, PA-28RT-201T, PA-28R-200, PA-28-201T.	The New Piper Aircraft, Inc.	2A13, Revision 47	CAR 3.
1 Piper	PA-30, PA-39, PA-40	The New Piper Aircraft, Inc.	A1EA, Revision 16	CAR 3.
1 Piper	PA-32-260, PA-32R-301 (SP), PA-32-300, PA-32R-301 (HP), PA-32S-300, PA-32R-301T, PA-32R-300, PA-32-301, PA-32RT-300, PA-32-301T, PA-32RT-300T, PA-32-301FT, PA-32-301XTC.	The New Piper Aircraft, Inc.	A3SO, Revision 29	CAR 3.
1 Piper	PA-34-200, PA-34-200T, PA-34-220T	The New Piper Aircraft, Inc.	A7SO, Revision 16	14 CFR PART 23.
1 Piper	PA-31P, PA-31T, PA-31T1, PA-31T2, PA-31T3, PA-31P-350.	The New Piper Aircraft, Inc.	A8EA, Revision 22	CAR 3.
1 Piper	PA-36-285, PA-36-300, PA-36-375	The New Piper Aircraft, Inc.	A9SO, Revision 9	14 CFR PART 23.
1 Piper	PA-36-285, PA-36-300, PA-36-375	The New Piper Aircraft, Inc.	A10SO, Revision 12	14 CFR PART 21/14 CFR PART 23.
Piper	PA-38-112	The New Piper Aircraft, Inc.	A18SO, Revision 4	14 CFR PART 23.
1 Piper	PA-44-180, PA-44-180T	The New Piper Aircraft, Inc.	A19SO, Revision 9	14 CFR PART 23.
1 Piper	PA-31, PA-31-300, PA-31-325, PA-31-350.	The New Piper Aircraft, Inc.	A20SO, Revision 10	CAR 3.
1 Piper	PA-42, PA-42-720, PA-42-1000	The New Piper Aircraft, Inc.	A23SO, Revision 17	14 CFR PART 23.
1 Piper	PA-46-310P, PA-46-350P, PA-46-500TP.	The New Piper Aircraft, Inc.	A25SO, Revision 14	14 CFR PART 23.
Piper	PA-46R-350T	Piper Aircraft, Inc	A25SO, Revision 16	14 CFR PART 23.
1 Tiger Aircraft LLC (American General).	AA-1, AA-1A, AA-1B, AA-1C	Tiger Aircraft LLC	A11EA, Revision 10	14 CFR PART 23.
1 Tiger Aircraft	AA-5, AA-5A, AA-5B, AG-5B	Tiger Aircraft LLC	A16EA, Revision 13	14 CFR PART 23.
1 Twin Commander	500, 500-A, 500-B, 500-U, 520, 560, 560-A, 560-E, 500-S.	Twin Commander Aircraft Corporation.	6A1, Revision 45	CAR 3.
1 Twin Commander	560-F, 681, 680, 690, 680E, 685, 680F, 690A, 720, 690B, 680FL, 690C, 680FL(P), 690D, 680T, 695, 680V, 695A, 680W, 695B.	Twin Commander Aircraft Corporation.	2A4, Revision 46	CAR 3.
1 Univair (Stinson)	108, 108-1, 108-2, 108-3, 108-5	Univair Aircraft Corporation.	A-767, Revision 27	CAR 3.
1 Univair	(ERCO) 415-D	Univair Aircraft Corporation.	A-787, Revision 33	CAR 3.
	(ERCO) E			
	(ERCO) G			
	(Forney) F-1			
	(Forney) F-1A			
	(Alon) A-2			
	(Alon) A2-A			
	(Mooney) M10			
1 Univair (Mooney)	(ERCO) 415-C, (ERCO) 415-CD	Univair Aircraft Corporation.	A-718, Revision 29	CAR 4a.

The following aircraft are certified in the restricted category:

LIST OF ALL AIRPLANE MODELS AND APPLICABLE TCDS

Make	Model	TC holder	TCDS	Certification basis
Air Tractor	AT-250, AT-300, AT-301, AT-302, AT-400, AT-400A.	Air Tractor, Inc	A9SW, Revision 12	14 CFR PART 23.
Air Tractor	AT-401, AT-401A, AT-401B, AT-402, AT-402A, AT-402B, AT-501, AT-502, AT-502A, AT-502B, AT-503, AT-503A.	Air Tractor, Inc	A17SW, Revision 10	14 CFR PART 23.
Air Tractor	AT-802A, AT-802, AT-602	Air Tractor, Inc	A19SW, Revision 4	14 CFR PART 23.
Allied Ag Cat	G-164, G-164A, G-164B, G-164B with 73", G-164B-15T, G-164B-34T, G-164B-20T, G-164C, G-164D, G-164D with 73" wing gap.	Allied Ag Cat Productions, Inc.	1A16, Revision 24	CAR 8.
Gippsland Aeronautics	GA200	Gippsland Aeronautics Pty. Ltd..	A00001LA, Revision 1	14 CFR PART 23.
2 Piper	PA-18A, PA-18A "135", PA-18A "150"	The New Piper Aircraft, Inc.	AR-7, Revision 11	CAR 8.
LAVIA S.A. (Piper)	PA-25, PA-25-235, PA-25-260	Latino Americana De Aviación (LAVIA) S.A.	2A10, Revision 24	CAR 8.
Thrush Aircraft, Inc. (Snow, Rockwell, Ayres).	S-2B, S-2C, 600-S2C	Thrush Aircraft, Inc	2A7, Revision 16	CAR 8.
Thrush Aircraft, Inc. (Snow, Rockwell, Ayres).	600 S-2D, S-2R, S2R-T34, S2R-T15, S2R-T11, S2R-R3S, S2R-R1340.	Thrush Aircraft, Inc	A3SW, Revision 18	CAR 3.
Thrush Aircraft, Inc. (Snow, Rockwell, Ayres).	600 S2D, S2R-R1340, S2R-G10, S-2R, S2R-R1820, S2R-G5, S2R-T34, S2R-T65, S2R-G1, S2R-T15, S2RHG-T65, S2RHG-T34, S2R-R3S, S2R-T45, S2R-T660, S2R-T11, S2R-G6.	Thrush Aircraft, Inc	A4SW, Revision 28	CAR 8.
Weatherly	620, 620TP, 620A, 620B, 620B-TG	Weatherly Aircraft Company.	A26WE, Revision 7	14 CFR PART 23.
3 Ximango	AMT-100, AMT-200, AMT-300, AMT-200S.	Aeromot-Industria Mecanico Metalugica Ltd.	TG00004AT, Revision 4	14 CFR PART 23.

Aircraft identified with a 1 have special conditions for AmSafe Aviation Inflatable Restraints published under Special Conditions 23-182-SC. Piper PA-18A, PA-18A "135" and PA-18A "150" (identified with a 2) are type certificated in Normal/Utility Category on TCDS 1A2 and in Restricted Category on TCDS AR-7. The same aircraft may be operated under either TCDS in accordance with the restrictions listed on TCDS AR-7.

Ximango (identified with a 3) is certificated in the Utility Category.

For all the models listed above, the certification basis also includes all exemptions, if any; equivalent level of safety findings, if any; and the other special conditions.

If the Administrator finds that the applicable airworthiness regulations (*i.e.*, part 23 as amended) do not contain adequate or appropriate safety standards for the AmSafe Aviation, inflatable restraint as installed on these models because of a novel or unusual design feature, special conditions are prescribed under the provisions of § 21.16.

Special conditions, as appropriate, as defined in § 11.19, are issued in accordance with § 11.38, and become part of the type certification basis in accordance with § 21.101.

Special conditions are initially applicable to the model for which they are issued. Should the applicant apply for a supplemental type certificate to modify any other model included on the same type certificate to incorporate the same novel or unusual design feature,

the special conditions would also apply to that model under the provisions of § 21.101.

Novel or Unusual Design Features

The various airplane models will incorporate the following novel or unusual design feature:

The AmSafe Aviation Inflatable Two-, Three-, Four-, or Five-Point Restraint Safety Belt with an Integrated Airbag Device.

The purpose of the airbag is to reduce the potential for injury in the event of an accident. In a severe impact, an airbag will deploy from the restraint, in a manner similar to an automotive airbag. The airbag will deploy between the head of the occupant and airplane interior structure. This will, therefore, provide some protection to the head of the occupant. The restraint will rely on sensors to electronically activate the inflator for deployment.

The Code of Federal Regulations state performance criteria for seats and restraints in an objective manner. However, none of these criteria are

adequate to address the specific issues raised concerning inflatable restraints. Therefore, the FAA has determined that, in addition to the requirements of part 21 and part 23, special conditions are needed to address the installation of this inflatable restraint.

Accordingly, these amended special conditions are adopted for the various airplane models equipped with the AmSafe Aviation, two-, three-, four-, or five-point inflatable restraint. Other conditions may be developed, as needed, based on further FAA review and discussions with the manufacturer and civil aviation authorities.

Applicability

As discussed above, these amended special conditions are applicable to the Approved Model List (AML) above. Should AmSafe Aviation apply at a later date for a supplemental type certificate to modify any other model included on the type certificates listed above to incorporate the same novel or unusual

design feature, the special conditions would apply to that model as well.

Conclusion

This action affects only certain novel or unusual design features on the previously identified airplane models. It is not a rule of general applicability, and it affects only the applicant who applied to the FAA for approval of these features on the airplane.

Under standard practice, the effective date of final special conditions would be 30 days after the date of publication in the **Federal Register**; however, as the certification date for these airplane models, as modified by AmSafe Aviation, is imminent, the FAA finds that good cause exists to make these amended special conditions effective upon issuance.

List of Subjects in 14 CFR Part 23

Aircraft, Aviation safety, Signs and symbols.

Citation

■ The authority citation for these amended special conditions is as follows:

Authority: 49 U.S.C. 106(g), 40113 and 44701; 14 CFR 21.16 and 21.101; and 14 CFR 11.38 and 11.19.

The Amended Special Conditions

The FAA has determined that this project will be accomplished on the basis of not lowering the current level of safety of the occupant restraint system for the airplane models listed in these special conditions. Accordingly, the FAA is issuing the following amended special conditions as part of the type certification basis for these models, as modified by AmSafe, Aviation.

Inflatable Two-, Three-, Four-, or Five-Point Restraint Safety Belt with an Integrated Airbag Device Installed in an Airplane Model.

1a. It must be shown that the inflatable restraint will provide restraint protection under the emergency landing conditions specified in the original certification basis of the airplane. Compliance will be demonstrated using the static test conditions specified in the original certification basis for each airplane.

1b. It must be shown that the crash sensor will trigger when exposed to a rapidly applied deceleration, like an actual emergency landing event. Therefore, compliance may be demonstrated using the deceleration pulse specified in § 23.562, which may be modified as follows:

I. The peak longitudinal deceleration may be reduced; however, the onset rate of the

deceleration must be equal to or greater than the emergency landing pulse identified in § 23.562.

II. The peak longitudinal deceleration must be above the deployment threshold of the sensor, and equal to or greater than the forward static design longitudinal load factor required by the original certification basis of the airplane.

2. The inflatable restraint must provide adequate protection for each occupant. In addition, unoccupied seats that have an active restraint must not constitute a hazard to any occupant.

3. The design must prevent the inflatable restraint from being incorrectly buckled and/or incorrectly installed such that the airbag would not properly deploy. Alternatively, it must be shown that such deployment is not hazardous to the occupant and will provide the required protection.

4. It must be shown that the inflatable restraint system is not susceptible to inadvertent deployment as a result of wear and tear or the inertial loads resulting from in-flight or ground maneuvers (including gusts and hard landings) that are likely to be experienced in service.

5. It must be extremely improbable for an inadvertent deployment of the restraint system to occur, or an inadvertent deployment must not impede the pilot's ability to maintain control of the airplane or cause an unsafe condition (or hazard to the airplane). In addition, a deployed inflatable restraint must be at least as strong as a Technical Standard Order (C22g or C114) restraint.

6. It must be shown that deployment of the inflatable restraint system is not hazardous to the occupant or will not result in injuries that could impede rapid egress. This assessment should include occupants whose restraints are loosely fastened.

7. It must be shown that an inadvertent deployment that could cause injury to a sitting person is improbable. In addition, the restraint must also provide suitable visual warnings that would alert rescue personnel to the presence of an inflatable restraint system.

8. It must be shown that the inflatable restraint will not impede rapid egress of the occupants 10 seconds after its deployment.

9. For the purposes of complying with HIRF and lightning requirements, the inflatable restraint system is considered a critical system since its deployment could have a hazardous effect on the airplane.

10. It must be shown that the inflatable restraints will not release hazardous quantities of gas or particulate matter into the cabin.

11. The inflatable restraint system installation must be protected from the effects of fire such that no hazard to occupants will result.

12. There must be a means to verify the integrity of the inflatable restraint activation system before each flight or it must be demonstrated to reliably operate between inspection intervals.

13. A life limit must be established for appropriate system components.

14. Qualification testing of the internal firing mechanism must be performed at vibration levels appropriate for a general aviation airplane.

Issued in Kansas City, Missouri on October 31, 2008.

James E. Jackson,

Acting Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. E8-26663 Filed 11-6-08; 8:45 am]

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DEPARTMENT OF HOMELAND SECURITY

U.S. Customs and Border Protection

19 CFR Part 102

[CBP Dec. 08-42]

Technical Corrections Relating to the Rules of Origin for Goods Imported Under the NAFTA and for Textile and Apparel Products

Correction

In rule document E8-25734 beginning on page 64518 in the issue of Thursday, October 30, 2008, make the following correction:

§102.21 [Corrected]

On page 64539, in §102.21, in the table, in the first column, in the first entry, "6209.20.1000...." should read "6209.20.1000-".

[FR Doc. Z8-25734 Filed 11-6-08; 8:45 am]

BILLING CODE 1505-01-D

DEPARTMENT OF LABOR

Mine Safety and Health Administration

30 CFR Parts 56, 57, and 71

RIN 1219-AB24

Asbestos Exposure Limit

AGENCY: Mine Safety and Health Administration, Labor.

ACTION: Final rule, technical amendment.