

under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption ADDRESSES.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

#### The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

##### § 39.13 [Amended]

2. Section 39.13 is amended by removing Amendment 39-11625 (65 FR 14209, March 16, 2000), and by adding a new airworthiness directive (AD), to read as follows:

**Eurocopter France:** Docket No. 2002-SW-32-AD. Supersedes AD 2000-05-15, Amendment 39-11625, Docket No. 99-SW-87-AD.

**Applicability:** Model AS355N helicopters, certificated in any category.

**Note 1:** This AD applies to each helicopter identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For helicopters that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (e) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent excessive starter-generator (S-G) vibration, which may lead to separation of an engine exhaust pipe ejector (ejector), impact with the main or tail rotor, and subsequent loss of control of the helicopter, accomplish the following:

(a) Before further flight and at or between 10 and 15 hours time-in-service (TIS), inspect the torque on each S-G attachment clamp (clamp). If the torque is not within tolerances provided in the maintenance manual, adjust the torque accordingly.

(b) Measure and record on a component history card or equivalent record the

vibration level for each S-G in accordance with the Accomplishment Instructions, paragraph 2.A.2., of Eurocopter France (ECF) Telex No. 01.00.45 R3, dated November 22, 2001 (Telex), as follows:

(1) For each S-G with less than 10 hours TIS since initial installation, before further flight, and at or between the hours TIS as shown in Table 1 of this AD:

TABLE 1.—S-G VIBRATION LEVEL MEASUREMENT INTERVALS

Hours TIS
A. 10 and 15.
B. 24 and 35.
C. 45 and 55.
D. 70 and 80.
E. 100 and 110.

(2) For each S-G with 10 hours or more TIS but less than 110 hours TIS since initial installation, begin and continue the vibration level measurements at or between the applicable hours TIS shown in Table 1 of this AD.

(3) For each S-G with more than 110 hours TIS since initial installation, measure the vibration level before further flight.

(c) After doing paragraph (b) of this AD, thereafter, at intervals not to exceed 110 hours TIS, measure the vibration level in accordance with paragraph 2.A.2. of the Telex.

(d) If the vibration level of an S-G is equal to or greater than 0.5 inches per second (IPS) (12.7 mm/s):

(1) Remove the S-G and repair or replace it with an airworthy S-G.

(2) Visually inspect the four ejector attachment lugs (lugs) and the two clamps for a crack in accordance with the Accomplishment Instructions, paragraph 2.B.3.b.1B), of the Telex.

(3) Inspect the two half-clamps for a crack.

(4) Remove the S-G to engine attachment flange (flange). Clean and inspect the flange for a crack in accordance with the Accomplishment Instructions, paragraph 2.B.3.b.1D) of the Telex.

(5) If a crack is found, before further flight, repair or replace the cracked part with an airworthy part in accordance with the Accomplishment Instructions, paragraph 2.B.3.b.3 of the Telex, except you are not required to report your findings to the manufacturer.

(e) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Regulations Group, Rotorcraft Directorate, FAA. Operators shall submit their requests through an FAA Principal Maintenance Inspector, who may concur or comment and then send it to the Manager, Regulations Group.

**Note 2:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Regulations Group.

(f) Special flight permits may be issued in accordance with 14 CFR 21.197 and 21.199 to operate the helicopter to a location where the requirements of this AD can be accomplished.

**Note 3:** The subject of this AD is addressed in Direction Generale De L'Aviation Civile (France) AD Nos. 1999-469-058(A)R1, dated August 9, 2000, and 1999-469-058(A)R2, dated January 9, 2002.

Issued in Fort Worth, Texas, on July 3, 2002.

**David A. Downey,**

*Manager, Rotorcraft Directorate, Aircraft Certification Service.*

[FR Doc. 02-17301 Filed 7-12-02; 8:45 am]

BILLING CODE 4910-13-P

#### DEPARTMENT OF TRANSPORTATION

#### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 2002-SW-06-AD]

RIN 2120-AA64

#### Airworthiness Directives; Agusta S.p.A. Model A109E Helicopters

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** This document proposes adopting a new airworthiness directive (AD) for Agusta S.p.A. (Agusta) Model A109E helicopters. This proposal would require establishing or reducing the life limits of various parts listed in the airworthiness limitations section (ALS) of the maintenance manual. This proposal is prompted by the results of fatigue tests and analysis to determine life limits for various parts. The actions specified by this proposed AD are intended to establish or reduce the life limits to prevent failure of specified parts and subsequent loss of control of the helicopter.

**DATES:** Comments must be received on or before September 13, 2002.

**ADDRESSES:** Submit comments in triplicate to the Federal Aviation Administration (FAA), Office of the Regional Counsel, Southwest Region, Attention: Rules Docket No. 2002-SW-06-AD, 2601 Meacham Blvd., Room 663, Fort Worth, Texas 76137. You may also send comments electronically to the Rules Docket at the following address: [9-asw-adcomments@faa.gov](mailto:9-asw-adcomments@faa.gov). Comments may be inspected at the Office of the Regional Counsel between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays.

#### FOR FURTHER INFORMATION CONTACT:

Carroll Wright, Aviation Safety Engineer, FAA, Rotorcraft Directorate, Regulations Group, Fort Worth, Texas 76193-0111, telephone (817) 222-5120, fax (817) 222-5961.

SUPPLEMENTARY INFORMATION:

Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications should identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments will be considered before taking action on the proposed rule. The proposals contained in this document may be changed in light of the comments received.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their mailed comments submitted in response to this proposal must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket No. 2002-SW-06-AD." The postcard will be date stamped and returned to the commenter.

Discussion

On September 7, 2001, Agusta sent the FAA a comparison between the May 1996 and July 2001 issues of the ALS of its maintenance manual and provided justification for the changes in fatigue lives of certain parts. The justification for changing the life limits was based on applying new fatigue life computations, a rescue hoist flight spectrum, a Category A training flight spectrum, a new operational limit, and an update of loads in-flight survey data.

This helicopter model is manufactured in Italy and is type certificated for operation in the United States under the provisions of 14 CFR 21.29 and the applicable bilateral

agreement. The FAA has reviewed all available information, and determined that AD action is necessary for products of this type design that are certificated for operation in the United States.

An unsafe condition is likely to exist or develop on other helicopters of the same type design registered in the United States if the proposed life limits are not followed. Therefore, the proposed AD would require establishing or reducing the life limits of specified parts of the main transmission assembly and supports, the tail rotor assemblies, the main rotor control bolt, and the fuselage left-hand elevator, and revising the ALS of the maintenance manual accordingly.

The FAA estimates that 31 helicopters of U.S. registry would be affected by this proposed AD. One copy of each of the 11 parts listed in Table 1 of this proposal would cost approximately \$41,294. Based on these figures, the total cost impact of the proposed AD on U.S. operators is estimated to be \$1,280,114, assuming that one copy of each part would be replaced on the entire fleet. There would be no additional labor costs as the parts would be replaced during the normal maintenance process.

The regulations proposed herein would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this proposal would not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by

contacting the Rules Docket at the location provided under the caption ADDRESSES.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:  
**Authority:** 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding a new airworthiness directive to read as follows:

**Agusta S.p.A.:** Docket No. 2002-SW-06-AD.

*Applicability:* Model A109E helicopters, certificated in any category.

**Note 1:** This AD applies to each helicopter identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For helicopters that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (c) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

*Compliance:* Required within 100 hours time-in-service (TIS), unless accomplished previously.

To prevent failure of specified parts of the main transmission assembly and supports, the tail rotor assemblies, the main rotor control bolt, or the fuselage left-hand elevator, and subsequent loss of control of the helicopter, accomplish the following:

- (a) Replace each part listed in Table 1 with an airworthy part on or before reaching the specified hours TIS as shown in Table 1 of this AD as follows:

TABLE 1

Part name	Part No.	Hours TIS
(1) Main transmission gear pinion .....	109-0403-05-111	6,100
(2) Main transmission gear driver .....	109-0403-04-3	8,300
(3) Main transmission shaft assembly .....	109-0405-76-107	25,000

TABLE 1—Continued

Part name	Part No.	Hours TIS
(4) Tail rotor retention strap assembly .....	109-8131-07-1	1,800
(5) Tail rotor hub assembly .....	109-0131-06-7	3,000
(6) Tail rotor 90-degree gearbox pinion gear .....	109-0433-01-107	6,100
(7) Tail rotor 90-degree gearbox crown gear .....	109-0443-01-103	11,700
(8) Main rotor control bolt .....	109-0110-90-103	5,000
(9) Fuselage left-hand elevator .....	109-0200-02-93	4,400
(10) Main transmission support aft rod .....	109-0325-03-113	35,000
(11) Main transmission support lower fitting .....	109-0325-08-1	30,000

(b) This AD revises the airworthiness limitations section of the maintenance manual by establishing or reducing the life limit as specified in Table 1 of this AD.

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Regulations Group, Rotorcraft Directorate, FAA. Operators shall submit their requests through an FAA Principal Maintenance Inspector, who may concur or comment and then send it to the Manager, Regulations Group.

**Note 2:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Regulations Group.

(d) Special flight permits will not be issued.

Issued in Fort Worth, Texas, on July 5, 2002.

**Larry M. Kelly,**

*Acting Manager, Rotorcraft Directorate,  
Aircraft Certification Service.*

[FR Doc. 02-17424 Filed 7-12-02; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 2002-CE-13-AD]

RIN 2120-AA64

#### Airworthiness Directives; Vulcanair S.p.A. P 68 Series Airplanes

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** This document proposes to adopt a new airworthiness directive (AD) that would apply to certain Vulcanair S.p.A. (Vulcanair) P 68 series airplanes. This proposed AD would require you to inspect the flight and

engine control systems to ensure that there is correct connecting bolt and linkage installation, no interference, and correct installation of certain components. The proposed AD would also require you to make any necessary adjustments and modify and install the split link and full travel limit assembly. This proposed AD is the result of mandatory continuing airworthiness information (MCAI) issued by the airworthiness authority for Italy. The actions specified by this proposed AD are intended to prevent failure of the primary flight control system caused by certain configurations. Such failure could lead to loss of aircraft flight control.

**DATES:** The Federal Aviation Administration (FAA) must receive any comments on this proposed rule on or before August 26, 2002.

**ADDRESSES:** Submit comments to FAA, Central Region, Office of the Regional Counsel, Attention: Rules Docket No. 2002-CE-13-AD, 901 Locust, Room 506, Kansas City, Missouri 64106. You may view any comments at this location between 8 a.m. and 4 p.m., Monday through Friday, except Federal holidays. You may also send comments electronically to the following address: [9-ACE-7-Docket@faa.gov](mailto:9-ACE-7-Docket@faa.gov). Comments sent electronically must contain "Docket No. 2002-CE-13-AD" in the subject line. If you send comments electronically as attached electronic files, the files must be formatted in Microsoft Word 97 for Windows or ASCII text.

You may get service information that applies to this proposed AD from Vulcanair S.p.A., Via G. Pascoli 7, 80026 Casoria (Naples) Italy, telephone: +39.081.5918111; facsimile: +39.081.5918172. You may also view this information at the Rules Docket at the address above.

#### FOR FURTHER INFORMATION CONTACT:

Doug Rudolph, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4059; facsimile: (816) 329-4090.

#### SUPPLEMENTARY INFORMATION:

##### Comments Invited

*How Do I Comment on This Proposed AD?*

The FAA invites comments on this proposed rule. You may submit whatever written data, views, or arguments you choose. You need to include the rule's docket number and submit your comments to the address specified under the caption **ADDRESSES**. We will consider all comments received on or before the closing date. We may amend this proposed rule in light of comments received. Factual information that supports your ideas and suggestions is extremely helpful in evaluating the effectiveness of this proposed AD action and determining whether we need to take additional rulemaking action.

*Are There Any Specific Portions of This Proposed AD I Should Pay Attention to?*

The FAA specifically invites comments on the overall regulatory, economic, environmental, and energy aspects of this proposed rule that might suggest a need to modify the rule. You may view all comments we receive before and after the closing date of the rule in the Rules Docket. We will file a report in the Rules Docket that summarizes each contact we have with the public that concerns the substantive parts of this proposed AD.

*How Can I be Sure FAA Receives My Comment?*

If you want FAA to acknowledge the receipt of your mailed comments, you must include a self-addressed, stamped postcard. On the postcard, write