

alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent ice accumulation on the airplane leading edges, which could result in reduced controllability of the airplane, accomplish the following:

Modification

(a) Within 1 year after the effective date of this AD, replace the pneumatic de-icing boot pressure indicator switch with a switch that activates the flight deck indicator light at 15 pounds per square inch gage, in accordance with a method approved by the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate.

Alternative Methods of Compliance

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

Special Flight Permits

(c) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Issued in Renton, Washington, on January 18, 2000.

Donald L. Riggin,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 99-NM-13-AD]

RIN 2120-AA64

Airworthiness Directives; Raytheon Model BAe 125-800A and BAe 125-800B, Model Hawker 800, and Model Hawker 800XP Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the superseding of an existing airworthiness directive (AD), applicable to certain Raytheon Model BAe 125-800A, Model Hawker 800, and Model Hawker 800XP series airplanes, that currently requires the filling of two tooling holes on the firewalls of the left and right engine pylons with firewall sealant. This action would require the sealing of all unused (open) tooling holes on the firewalls of the left and right engine pylons, and would expand the applicability to include additional airplanes. This proposal is prompted by reports of additional unused (open) tooling holes, found at locations other than those currently addressed. The actions specified by the proposed AD are intended to prevent an engine fire from moving to the fuselage and to the lines that carry flammable fluid that are located inboard of the firewall.

DATES: Comments must be received by February 23, 2000.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 99-NM-13-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056. Comments may be inspected at this location between 9:00 a.m. and 3:00 p.m., Monday through Friday, except Federal holidays.

The service information referenced in the proposed rule may be obtained from Raytheon Aircraft Company, Manager Service Engineering, Hawker Customer Support Department, P.O. Box 85, Wichita, Kansas, 67201-0085. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Small Airplane Directorate, Wichita Aircraft Certification Office, 1801 Airport Road, Room 100, Mid-Continent Airport, Wichita, Kansas.

FOR FURTHER INFORMATION CONTACT: Jeffrey A. Pretz, Aerospace Engineer, Systems and Propulsion Branch, ACE-116W, FAA, Small Airplane Directorate, Wichita Aircraft Certification Office, 1801 Airport Road, Room 100, Mid-Continent Airport, Wichita, Kansas 67209; telephone (316) 946-4153; fax (316) 946-4407.

SUPPLEMENTARY INFORMATION:

Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such

written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this notice may be changed in light of the comments received.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this notice must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 99-NM-13-AD." The postcard will be date stamped and returned to the commenter.

Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 99-NM-13-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056.

Discussion

On November 22, 1996, the FAA issued AD 96-24-16, amendment 39-9840 (61 FR 66878, December 19, 1996), applicable to certain Raytheon Model BAe 125-800A, Model Hawker 800, and Model Hawker 800XP series airplanes, to require the filling of two tooling holes on the firewalls of the left and right engine pylons with firewall sealant. That action was prompted by notification from the manufacturer that these holes were not sealed during production. The requirements of that AD are intended to prevent an engine fire from moving to the fuselage and to the lines that carry flammable fluid that are located inboard of the firewall.

Actions Since Issuance of Previous Rule

Since the issuance of that AD, reports have been received of airplanes with additional unused tooling holes, at locations other than those addressed in AD 96-24-16, on the left and right engine pylon firewalls, which may permit the passage of flames to the

structure and flammable fluids inboard of the engine pylon firewall.

Explanation of Relevant Service Information

Raytheon has issued Service Bulletin SB.54-1-3815B, Revision 1, dated May 1998, which describes procedures for the sealing of all unused tooling holes on the firewalls of the left and right engine pylons. Revision 1 of the service bulletin was issued to address additional unused engine pylon firewall tooling holes at locations other than those indicated in the original service bulletin, dated March 26, 1996. Accomplishment of the actions specified in Raytheon Service Bulletin SB.54-1-3815B, Revision 1, is intended to adequately address the identified unsafe condition.

Explanation of Requirements of Proposed Rule

Since an unsafe condition has been identified that is likely to exist or develop on other airplanes of the same type design, the proposed AD would supersede AD 96-24-16 to require the sealing of all unused tooling holes on the firewalls of the left and right engine pylons. The actions would be required to be accomplished in accordance with Raytheon Service Bulletin SB.54-1-3815B, Revision 1, described previously.

Expanded Applicability

The applicability of the proposed AD has been expanded to include Model BAe 125-800B series airplanes, which have received FAA type certification and have a similar design to airplanes subject to the requirements of AD 96-24-16.

Cost Impact

There are approximately 340 Model BAe 125-800A and BAe 125-800B, Model Hawker 800, and Model Hawker 800XP series airplanes of the affected design in the worldwide fleet. The FAA estimates that 221 airplanes of U.S. registry would be affected by this proposed AD.

The actions that are currently required by AD 96-24-16, and retained in this AD, take approximately 2 work hours per airplane to accomplish, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of the currently required actions on U.S. operators is estimated to be \$120 per airplane.

The new actions that are proposed in this AD action would take approximately 2 work hours per airplane to accomplish, at an average labor rate of \$60 per work hour. Based

on these figures, the cost impact of the proposed requirements of this AD on U.S. operators is estimated to be \$26,520, or \$120 per airplane.

The cost impact figures discussed above are based on assumptions that no operator has yet accomplished any of the current or proposed requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

Regulatory Impact

The regulations proposed herein would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this proposal would not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by removing amendment 39-9840 (61 FR 66878, December 19, 1996), and by adding a new airworthiness directive (AD), to read as follows:

Raytheon Aircraft Co. (Formerly Beech):

Docket 99-NM-13-AD. Supersedes AD 96-24-16, Amendment 39-9840.

Applicability: Model BAe 125-800A and BAe 125-800B, Model Hawker 800, and Model Hawker 800XP series airplanes; as listed in Raytheon Service Bulletin SB.54-1-3815B, Revision 1, dated May 1998; certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (c) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent an engine fire from moving to the fuselage and to the lines that carry flammable fluid that are located inboard of the firewall, accomplish the following:

Restatement of Requirements of AD 96-24-16

(a) For airplanes identified in AD 96-24-16, amendment 39-9840: Within 6 months after January 27, 1997 (the effective date of AD 96-24-16), fill the two, unused tooling holes in the firewalls of the left and right engine pylons, in accordance with Raytheon Service Bulletin SB.54-1-3815B, dated March 26, 1996, or Raytheon Service Bulletin SB.54-1-3815B, Revision 1, dated May 1998. After the effective date of this AD, only Revision 1 of this service bulletin shall be used.

New Requirements of This AD

(b) For all airplanes: Within 6 months after the effective date of this AD, fill all unused tooling holes in the left and right engine pylon firewalls with firewall sealant, in accordance with Raytheon Service Bulletin SB.54-1-3815B, Revision 1, dated May 1998.

Alternative Methods of Compliance

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Wichita Aircraft Certification Office (ACO), ACE-116W, FAA, Small Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Wichita ACO.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Wichita ACO.

Special Flight Permits

(d) Special flight permits may be issued in accordance with sections §§ 21.197 and

21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Issued in Renton, Washington, on January 18, 2000.

Donald L. Riggin,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 99-NE-40-AD]

RIN 2120-AA64

Airworthiness Directives; CFM International CFM56-2, -2A, -2B, -3, -3B, -3C, -5, -5B, -5C Series Turbofan Engines

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the adoption of a new airworthiness directive (AD) that is applicable to CFM International (CFMI) CFM56-2, -2A, -2B, -3, -3B, -3C, -5, -5B, -5C series turbofan engines. This proposal would require initial and repetitive visual inspections of the fuel pump filter cover helicoil inserts and bolts for damage, and, if necessary, repair or replacement with serviceable parts. This proposal also would require the installation of new fuel pumps that incorporate an improved filter cover retention design (D-bolts) as terminating action to the inspections. This proposal is prompted by reports that fuel pump filter cover helicoil inserts have loosened or pulled out. The actions specified by the proposed AD are intended to prevent fuel leakage from between the fuel pump filter cover and gear housing, which could result in an engine fire and damage to the airplane.

DATES: Comments must be received by February 23, 2000.

ADDRESSES: Submit comments to the Federal Aviation Administration (FAA), New England Region, Office of the Regional Counsel, Attention: Rules Docket No. 99-NE-40-AD, 12 New England Executive Park, Burlington, MA 01803-5299. Comments may also be sent via the Internet using the following address: "9-ane-adcomment@faa.gov". Comments sent via the Internet must contain the docket number in the

subject line. Comments may be inspected at this location between 8:00 a.m. and 4:30 p.m., Monday through Friday, except Federal holidays.

The service information referenced in the proposed rule may be obtained from CFM International, Technical Publications Department, 1 Neumann Way, Cincinnati, OH 45215; telephone (513) 552-2981, fax (513) 552-2816. This information may be examined at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA.

FOR FURTHER INFORMATION CONTACT: James Rosa, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803-5299; telephone (781) 238-7152, fax (781) 238-7199.

SUPPLEMENTARY INFORMATION:

Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications should identify the Rules Docket number and be submitted to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this notice may be changed in light of the comments received.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this notice must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 99-NE-40-AD." The postcard will be date stamped and returned to the commenter.

Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA, New England Region, Office of the Regional Counsel, Attention: Rules Docket No. 99-NE-40-AD, 12 New England Executive Park, Burlington, MA 01803-5299.

Discussion

The Federal Aviation Administration (FAA) has received reports of fuel leakage from between the fuel pump filter cover and gear housing on CFM International (CFMI) CFM56-2, -2A, -2B, -3, -3B, -3C, -5, -5B, -5C series turbofan engines.

Investigation

The investigation revealed that fuel pump filter cover helicoil inserts have loosened or pulled out. The FAA has identified excessive torque during the installation of fuel filter cover bolts as the cause.

Improved Filter Cover

The manufacturer has introduced an improved filter cover retention design using a captured D-bolt and special locking nut that reduces the potential for filter cover bolt over torque damage and loss of bolted joint strength.

Unsafe Condition

This condition, if not corrected, could result in fuel leakage from between the fuel pump filter cover and gear housing, which could result in an engine fire and damage to the airplane.

Service Information

The FAA has reviewed and approved the technical contents of the following CFMI Service Bulletins (SBs), that describe procedures for visual inspections of inserts and bolts for damage, and provide criteria for rejection of hardware and necessary replacement with serviceable parts: CFM56-2 SB 73-110, Revision 2, dated April 29, 1999; CFM56

The FAA has also reviewed and approved the technical contents of the following CFMI SBs, that describe procedures for removal and replacement of fuel pumps (D-bolt fix): CFM56-2 SB 73-A113, dated August 17, 1999; CFM56-2A SB 73-A058, dated August 17, 1999; CFM56-2B SB 73-A079, dated August 17, 1999; CFM56-3/3B/3C SB 73-A129, dated August 17, 1999; CFM56-5 SB 73-A143, dated June 18, 1999; CFM56-5B SB 73-A062, dated June 18, 1999; CFM56-5C SB 73-A078, dated June 21, 1999.

Differences Between Service Bulletins and This AD

The referenced SBs describe a one-time inspection. This AD requires repetitive inspections at every filter change.

Proposed Inspections and Repair or Replacement

Since an unsafe condition has been identified that is likely to exist or