

operator has yet accomplished any of the proposed requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

### Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action: (1) Is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

### Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

### PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### **§ 39.13 [Amended]**

2. Section 39.13 is amended by adding the following new airworthiness directive:

**2000-01-04 Saab Aircraft AB:** Amendment 39-11501. Docket 99-NM-244-AD.

**Applicability:** All Model SAAB 2000 series airplanes, certificated in any category.

**Note 1:** This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the

requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent improper illumination of the ground under the service door due to incorrect installation of the aft exterior light, which could result in injury to the passengers or crew members during an emergency evacuation, accomplish the following:

(a) Within 4 months after the effective date of this AD, perform a one-time general visual inspection of the aft exterior light to verify proper orientation, in accordance with Saab Service Bulletin 2000-33-016, dated April 21, 1999.

(1) If the aft exterior light is correctly installed, as specified in the service bulletin, reinstall the lens in accordance with the service bulletin.

(2) If the aft exterior light is incorrectly installed, as specified in the service bulletin, prior to further flight, correct the orientation of the aft exterior light in accordance with the service bulletin.

**Note 2:** For the purposes of this AD, a general visual inspection is defined as: "A visual examination of an interior or exterior area, installation, or assembly to detect obvious damage, failure, or irregularity. This level of inspection is made under normally available lighting conditions such as daylight, hangar lighting, flashlight, or drop-light, and may require removal or opening of access panels or doors. Stands, ladders, or platforms may be required to gain proximity to the area being checked."

### Alternative Methods of Compliance

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

**Note 3:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

### Special Flight Permits

(c) Special flight permits may be issued in accordance with §§ 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

### Incorporation by Reference

(d) The actions shall be done in accordance with Saab Service Bulletin 2000-33-016,

dated April 21, 1999. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Saab Aircraft AB, SAAB Aircraft Product Support, S-581.88, Linköping, Sweden. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**Note 4:** The subject of this AD is addressed in Swedish airworthiness directive No. 1-140, dated April 21, 1999.

(e) This amendment becomes effective on February 16, 2000.

Issued in Renton, Washington, on January 4, 2000.

**Donald L. Riggins,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*  
[FR Doc. 00-501 Filed 1-11-00; 8:45 am]

**BILLING CODE 4910-13-U**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 99-NM-177-AD; Amendment 39-11505; AD 2000-01-08]

**RIN 2120-AA64**

### Airworthiness Directives; British Aerospace BAe Model ATP Airplanes

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to all British Aerospace BAe Model ATP airplanes, that requires a one-time inspection of the orientation of certain bolts of the rudder standby control system (SCS), and reinstallation of the bolts, if necessary. This amendment is prompted by issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by this AD are intended to prevent uncommanded engagement of the rudder SCS, which could result in reduced controllability of the airplane.

**DATES:** Effective February 16, 2000.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of February 16, 2000.

**ADDRESSES:** The service information referenced in this AD may be obtained from British Aerospace Regional Aircraft American Support, 13850 Mclearen Road, Herndon, Virginia

20171. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:**

Norman B. Martenson, Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2110; fax (425) 227-1149.

**SUPPLEMENTARY INFORMATION:** A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to all British Aerospace BAe Model ATP airplanes was published in the **Federal Register** on November 8, 1999 (64 FR 60748). That action proposed to require a one-time inspection of the orientation of certain bolts of the rudder standby control system (SCS), and reinstallation of the bolts, if necessary.

**Comments**

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were submitted in response to the proposal or the FAA's determination of the cost to the public.

**Conclusion**

The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

**Cost Impact**

The FAA estimates that 10 airplanes of U.S. registry will be affected by this AD, that it will take approximately 3 work hours per airplane to accomplish the required inspection, and that the average labor rate is \$60 per work hour. Based on these figures, the cost impact of the AD on U.S. operators is estimated to be \$1,800, or \$180 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the proposed requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

**Regulatory Impact**

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not

have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

**Adoption of the Amendment**

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

2. Section 39.13 is amended by adding the following new airworthiness directive:

**2000-01-08 British Aerospace Regional Aircraft [Formerly Jetstream Aircraft Limited; British Aerospace (Commercial Aircraft) Limited]:** Amendment 39-11505. Docket 99-NM-177-AD.

*Applicability:* All BAe Model ATP airplanes, certificated in any category.

**Note 1:** This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

*Compliance:* Required as indicated, unless accomplished previously.

To prevent uncommanded engagement of the rudder standby control system (SCS), accomplish the following:

(a) Within one year after the effective date of this AD, perform a one-time general visual inspection of the orientation of the bolts in the rudder SCS, in accordance with British Aerospace Service Bulletin ATP-27-86, dated May 15, 1999. If any bolt is incorrectly installed, as specified by Figure 1 of the service bulletin, prior to further flight, remove and reinstall the bolt in accordance with the service bulletin.

**Note 2:** For the purposes of this AD, a general visual inspection is defined as: "A visual examination of an interior or exterior area, installation, or assembly to detect obvious damage, failure, or irregularity. This level of inspection is made under normally available lighting conditions such as daylight, hangar lighting, flashlight, or drop-light, and may require removal or opening of access panels or doors. Stands, ladders, or platforms may be required to gain proximity to the area being checked."

**Alternative Methods of Compliance**

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

**Note 3:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

**Special Flight Permits**

(c) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

**Incorporation by Reference**

(d) The actions shall be done in accordance with British Aerospace Service Bulletin ATP-27-86, dated May 15, 1999. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from British Aerospace Regional Aircraft American Support, 13850 Mclearen Road, Herndon, Virginia 20171. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**Note 4:** The subject of this AD is addressed in British airworthiness directive 005-05-99.

(e) This amendment becomes effective on February 16, 2000.

Issued in Renton, Washington, on January 4, 2000.

**Donald L. Riggin,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 00-500 Filed 1-11-00; 8:45 am]

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## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 99-NM-357-AD; Amendment 39-11504; AD 2000-01-07]

RIN 2120-AA64

#### **Airworthiness Directives; Bombardier Model DHC-8-100, -200, and -300 Series Airplanes**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule; request for comments.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD) that is applicable to certain Bombardier Model DHC-8-100, -200, and -300 series airplanes. This action requires the removal and testing of sections of bonded skin from the upper and lower skin panels of the horizontal stabilizer, repair of those areas, and follow-on corrective actions, if necessary. This amendment is prompted by issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified in this AD are intended to prevent reduced strength capability and consequent failure of the horizontal stabilizer, which could result in loss of controllability of the airplane.

**DATES:** Effective January 27, 2000.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of January 27, 2000.

Comments for inclusion in the Rules Docket must be received on or before February 11, 2000.

**ADDRESSES:** Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 99-NM-357-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056.

The service information referenced in this AD may be obtained from Bombardier, Inc., Bombardier Regional Aircraft Division, Garratt Boulevard, Downsview, Ontario M3K 1Y5, Canada. This information may be examined at

the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Engine and Propeller Directorate, New York Aircraft Certification Office, 10 Fifth Street, Third Floor, Valley Stream, New York; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

#### **FOR FURTHER INFORMATION CONTACT:**

Serge Napoleon, Aerospace Engineer, Airframe and Propulsion Branch, ANE-171, FAA, Engine and Propeller Directorate, New York Aircraft Certification Office, 10 Fifth Street, Third Floor, Valley Stream, New York 11581; telephone (516) 256-7512; fax (516) 568-2716.

**SUPPLEMENTARY INFORMATION:** Transport Canada Civil Aviation (TCCA), which is the airworthiness authority for Canada, notified the FAA that an unsafe condition may exist on certain Bombardier Model DHC-8-100, -200, and -300 series airplanes. TCCA advises that it received results of previous inspections to detect disbonding of the upper and lower skin panels of the horizontal stabilizer. Results of these inspections indicate that certain doublers and stringers were disbonded from their skin panels. The subject doublers and stringers are installed on the upper and lower skin panels of the horizontal stabilizer at stations Yh6.15-139.00 between the front spar and rear spar. The cause of such disbonding has been attributed to an improper bonding process of the skin panels of the doublers and stringers during manufacturing. This condition, if not corrected, could result in reduced strength capability and consequent failure of horizontal stabilizer, which could further result in loss of controllability of the airplane.

TCCA has recommended further testing on sample coupons of the sections of skin panels to determine the bonding integrity of the skin panels. A destructive peel test would be conducted on the removed sections of bonded skin (test coupons) by the manufacturer. This type of testing would determine how much force is necessary in order to separate the bonded layers of the skin panel.

#### **Explanation of Relevant Service Information**

The manufacturer has issued Bombardier Service Bulletin S.B. 8-55-24, dated February 26, 1998, which describes procedures for replacement of the horizontal stabilizer with a new horizontal stabilizer.

Accomplishment of the test on the sample coupons and, if necessary,

accomplishment of the actions specified in the service bulletin is intended to adequately address the identified unsafe condition. TCCA approved this service bulletin and issued Canadian airworthiness directive CF-99-15R1, dated December 6, 1999, in order to assure the continued airworthiness of these airplanes in Canada.

#### **Related Rulemaking**

On November 16, 1998, the FAA issued AD 98-24-18, amendment 39-10903 (63 FR 64609, November 23, 1998), applicable to certain Bombardier Model DHC-8-100 and -300 series airplanes equipped with Canadian Aviation Products (CAP) horizontal stabilizers having serial numbers 003 through 214, that requires repetitive ultrasonic inspections to detect disbonding of the upper and lower skin panels of the horizontal stabilizer, and repair, if necessary. Based on the information provided by TCCA, the FAA has determined that an additional disbonding test of the skin panels is needed on airplanes equipped with CAP horizontal stabilizers having serial numbers 003 through 050, the repetitive ultrasonic inspections required by AD 98-24-18 may not be adequate to detect disbonding of the upper and lower skin panels of the horizontal stabilizer, and, therefore, may not be providing an adequate level of safety for the transport airplane fleet. This AD affects the requirements of that AD. For certain airplanes, the corrective actions of this AD eliminate the need for the repetitive inspections required by AD 98-24-18.

#### **FAA's Conclusions**

This airplane model is manufactured in Canada and is type certificated for operation in the United States under the provisions of section 21.29 of the Federal Aviation Regulations (14 CFR 21.29) and the applicable bilateral airworthiness agreement. Pursuant to this bilateral airworthiness agreement, TCCA has kept the FAA informed of the situation described above. The FAA has examined the findings of TCCA, reviewed all available information, and determined that AD action is necessary for products of this type design that are certificated for operation in the United States.

#### **Explanation of Requirements of Rule**

Since an unsafe condition has been identified that is likely to exist or develop on other airplanes of the same type design registered in the United States, this AD is being issued to prevent reduced strength capability of the horizontal stabilizer and consequent failure of the horizontal stabilizer,