

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

Fokker: Docket 97–NM–249–AD.

Applicability: Model F28 Mark 0070 and Mark 0100 series airplanes; as listed in Fokker Service Bulletin SBF100–53–084, dated July 6, 1996; if equipped with any corrugated seal having part number (P/N) BE20061 (Rolls-Royce P/N 3405891); certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been otherwise modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (e) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent the leakage of hot air from the corrugated seals of low- and high-pressure check valves located in the stubwings, which could result in heat damage to the fuselage skin and stubwing structure and consequent reduced structural integrity, accomplish the following:

(a) Within 3,000 flight hours or 12 months after the effective date of this AD, whichever occurs first, perform a one-time visual inspection of the fuselage skin in the left- and right-hand stubwings to detect heat damage; in accordance with Part 2 of the Accomplishment Instructions of Fokker Service Bulletin SBF100–53–084, dated July 6, 1996.

(b) If no heat damage is found during the inspection required by paragraph (a) of this AD, prior to further flight, perform a leak test of each corrugated seal at the 7th stage low-pressure and 12th stage high-pressure check valves of the left- and right-hand bleed air systems, in accordance with Part 3 of the Accomplishment Instructions of Fokker Service Bulletin SBF100–53–084, dated July 6, 1996.

(1) If any leakage is found at a seal, prior to further flight, replace that seal with a new improved seal having part number EU15969, in accordance with the Accomplishment Instructions of Fokker Service Bulletin SBF100–36–026, Revision 1, dated July 6, 1996.

(2) If no leakage is found at a seal, perform an additional leak test of that seal within 250 flight hours after the initial test.

(i) If no leakage is found during the additional test of the seal, within 3,000 flight hours after the additional test, replace the seal with an improved seal having P/N EU15969, in accordance with the Accomplishment Instructions of Fokker Service Bulletin SBF100–36–026, Revision 1, dated July 6, 1996.

(ii) If any leakage is found during the additional test of the seal, prior to further flight, accomplish paragraphs (b)(2)(ii)(A) and (b)(2)(ii)(B) of this AD.

(A) Replace the seal with a new improved seal having P/N EU15969, in accordance with the Accomplishment Instructions of Fokker Service Bulletin SBF100–36–026, Revision 1, dated July 6, 1996.

(B) Inspect the fuselage skin in the applicable left- or right-hand stubwing to detect heat damage, in accordance with Part 2 of the Accomplishment Instructions of Fokker Service Bulletin SBF100–53–084, dated July 6, 1996.

(c) If any heat damage is found during the inspection required by paragraph (a) or paragraph (b)(2)(ii)(B) of this AD, prior to further flight, perform a detailed inspection of the fuselage skin and stubwing structure to detect the extent of heat damage, in accordance with Parts 4 and 5 of the Accomplishment Instructions of Fokker Service Bulletin SBF100–53–084, dated July 6, 1996; and accomplish paragraphs (c)(1) and (c)(2) of this AD.

(1) Repair the affected structure, in accordance with Part 6 of the Accomplishment Instructions of Fokker Service Bulletin SBF100–53–084, dated July 6, 1996. And

(2) Replace all corrugated seals having P/N BE20061 (Rolls-Royce P/N 3405891) at the 7th stage low-pressure and 12th stage high-pressure check valves of the left- and right-hand bleed air systems with new improved corrugated seals having P/N EU15969, in accordance with the Accomplishment Instructions of Fokker Service Bulletin SBF100–36–026, Revision 1, dated July 6, 1996.

(d) As of the effective date of this AD, no person shall install a corrugated seal having P/N BE20061 (Rolls-Royce P/N 3405891) on any airplane.

(e) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM–116, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM–116.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM–116.

(f) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Note 3: The subject of this AD is addressed in Dutch airworthiness directive BLA 1995–076/2 (A), dated August 30, 1996.

Issued in Renton, Washington, on November 20, 1997.

Stewart R. Miller,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 97–31159 Filed 11–26–97; 8:45 am]

BILLING CODE 4910–13–U

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 97–NM–264–AD]

RIN 2120–AA64

Airworthiness Directives; Fokker Model F28 Mark 0070 and Mark 0100 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the adoption of a new airworthiness directive (AD) that is applicable to all Fokker Model F28 Mark 0070 and Mark 0100 series airplanes. This proposal would require a one-time visual inspection to detect cracking of the brake torque tube lever, and corrective action, if necessary. This proposal is prompted by the issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by the proposed AD are intended to prevent failure of the brake torque tube lever, which could result in a disconnection between the brake pedal and brake system, and consequent reduced directional controllability of the airplane during landing.

DATES: Comments must be received by December 29, 1997.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM–103, Attention: Rules Docket No. 97–NM–264–AD, 1601 Lind Avenue, SW., Renton, Washington 98055–4056. Comments may be inspected at this location between 9:00 a.m. and 3:00 p.m., Monday through Friday, except Federal holidays.

The service information referenced in the proposed rule may be obtained from Fokker Service B.V., Technical Support Department, P.O. Box 75047, 1117 ZN Schiphol Airport, the Netherlands. This information may be examined at the

FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington.

FOR FURTHER INFORMATION CONTACT: International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2110; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION:

Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this notice may be changed in light of the comments received.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this notice must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 97-NM-264-AD." The postcard will be date stamped and returned to the commenter.

Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Transport Airplane Directorate, ANM-103, Attention: Rules Docket No. 97-NM-264-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056.

Discussion

The Rijksluchtvaartdienst (RLD), which is the airworthiness authority for the Netherlands, notified the FAA that an unsafe condition may exist on all Fokker Model F28 Mark 0070 and Mark 0100 series airplanes. The RLD advises that it has received a report of the failure of a brake torque tube lever, which occurred while the operator of a Fokker Model F28 Mark 0100 series airplane was attempting to set the

parking brake. Subsequent inspection during routine maintenance on two other airplanes in the fleet revealed small cracks in the same lever. Such cracking, if not corrected, could result in failure of the captain's left-hand brake torque tube lever, which could result in a disconnection between the captain's left-hand brake pedal and left-hand brake system, and consequent reduced directional controllability of the airplane during landing.

Explanation of Relevant Service Information

Fokker has issued Service Bulletin SBF100-32-108, dated February 7, 1997, which describes procedures for a one-time visual inspection using a mirror or borescope to detect cracking of the captain's left-hand brake torque tube lever. The service bulletin also describes procedures for replacement of any cracked lever with a new or serviceable lever, or, as an alternative, replacement of the entire brake torque tube assembly with a new or serviceable assembly. The RLD classified this service bulletin as mandatory and issued Dutch airworthiness directive 1997-025 (A), dated February 28, 1997, in order to assure the continued airworthiness of these airplanes in the Netherlands.

FAA's Conclusions

These airplane models are manufactured in the Netherlands and are type certificated for operation in the United States under the provisions of section 21.29 of the Federal Aviation Regulations (14 CFR 21.29) and the applicable bilateral airworthiness agreement. Pursuant to this bilateral airworthiness agreement, the RLD has kept the FAA informed of the situation described above. The FAA has examined the findings of the RLD, reviewed all available information, and determined that AD action is necessary for products of this type design that are certificated for operation in the United States.

Explanation of Requirements of Proposed Rule

Since an unsafe condition has been identified that is likely to exist or develop on other airplanes of the same type design registered in the United States, the proposed AD would require accomplishment of the actions specified in the service bulletin described previously.

Interim Action

This is considered to be interim action until final action is identified, at which time the FAA may consider further rulemaking.

Cost Impact

The FAA estimates that 131 Fokker Model F28 Mark 0070 and Mark 0100 series airplanes of U.S. registry would be affected by this proposed AD, that it would take approximately 3 work hours per airplane to accomplish the inspection proposed by this AD, and that the average labor rate is \$60 per work hour. Based on these figures, the cost impact of the proposed inspection of this AD on U.S. operators is estimated to be \$23,580, or \$180 per airplane.

The cost impact figures discussed above are based on assumptions that no operator has yet accomplished any of the proposed requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

Regulatory Impact

The regulations proposed herein would not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this proposal would not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption ADDRESSES.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

Fokker: Docket 97–NM–264–AD.

Applicability: All Model F28 Mark 0070 and Mark 0100 series airplanes, certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent failure of the captain's left-hand brake torque tube lever, which could result in a disconnection between the captain's left-hand brake pedal and left-hand brake system, and consequent reduced directional controllability of the airplane during landing, accomplish the following:

(a) Perform a one-time visual inspection using a mirror or borescope to detect cracking of the brake torque tube lever having part number (P/N) D75669–001, in accordance with Fokker Service Bulletin SBF100–32–108, dated February 7, 1997, at the time specified in paragraph (a)(1) or (a)(2), as applicable, of this AD. If any crack is detected, prior to further flight, replace either the lever or the entire assembly with a new or serviceable component, in accordance with the Accomplishment Instructions of the service bulletin.

(1) For airplanes that have accumulated 15,000 or more total flight cycles as of the effective date of this AD: Inspect within 30 days after the effective date of this AD.

(2) For airplanes that have accumulated fewer than 15,000 total flight cycles as of the effective date of this AD: Inspect prior to the accumulation of 10,000 total flight cycles, or within 2 months after the effective date of this AD, whichever occurs later.

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM–116, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM–116.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM–116.

(c) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Note 3: The subject of this AD is addressed in Dutch airworthiness directive 1997–025 (A), dated February 28, 1997.

Issued in Renton, Washington, on November 20, 1997.

Stewart R. Miller,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 97–31161 Filed 11–26–97; 8:45 am]

BILLING CODE 4910–13–U

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 97–NM–188–AD]

RIN 2120–AA64

Airworthiness Directives; Airbus Model A300, A310, and A300–600 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the adoption of a new airworthiness directive (AD) that is applicable to all Airbus Model A300, A310, and A300–600 series airplanes. For certain airplanes, this proposal would require replacing the bearings of the throttle control levers with new sealed bearings. For certain other airplanes, this proposal would require replacing the throttle control assemblies with new assemblies. This proposal is prompted by the issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by the proposed AD are intended to prevent asymmetric engine thrust on the airplane when the autothrottle is engaged, which could result in roll and yaw disturbances, and consequent reduced controllability of the airplane.

DATES: Comments must be received by December 29, 1997.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM–103, Attention: Rules Docket No. 97–NM–

188–AD, 1601 Lind Avenue, SW., Renton, Washington 98055–4056. Comments may be inspected at this location between 9:00 a.m. and 3:00 p.m., Monday through Friday, except Federal holidays.

The service information referenced in the proposed rule may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington.

FOR FURTHER INFORMATION CONTACT: International Branch, ANM–116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055–4556, telephone (425) 227–2110; fax (425) 227–1149.

SUPPLEMENTARY INFORMATION:

Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this notice may be changed in light of the comments received.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this notice must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 97–NM–188–AD." The postcard will be date stamped and returned to the commenter.

Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Transport Airplane Directorate, ANM–103, Attention: Rules Docket No. 97–NM–188–AD, 1601 Lind Avenue, SW., Renton, Washington 98055–4056.